



US009249727B2

(12) **United States Patent**
Matos

(10) **Patent No.:** **US 9,249,727 B2**
(45) **Date of Patent:** **Feb. 2, 2016**

(54) **RETRACTABLE BIRD AND DEBRIS
DEFLECTOR FOR AN AIRCRAFT JET
ENGINE**

(71) Applicant: **Jeffrey A. Matos**, New Rochelle, NY
(US)

(72) Inventor: **Jeffrey A. Matos**, New Rochelle, NY
(US)

(*) Notice: Subject to any disclaimer, the term of this
patent is extended or adjusted under 35
U.S.C. 154(b) by 494 days.

(21) Appl. No.: **13/799,396**

(22) Filed: **Mar. 13, 2013**

(65) **Prior Publication Data**

US 2013/0213003 A1 Aug. 22, 2013

Related U.S. Application Data

(63) Continuation-in-part of application No. 12/689,554,
filed on Jan. 19, 2010, now Pat. No. 8,429,890.

(60) Provisional application No. 61/205,381, filed on Jan.
16, 2009, provisional application No. 61/205,785,
filed on Jan. 22, 2009.

(51) **Int. Cl.**
F02C 7/00 (2006.01)
B64D 33/02 (2006.01)
F02C 7/055 (2006.01)

(52) **U.S. Cl.**
CPC . **F02C 7/00** (2013.01); **B64D 33/02** (2013.01);
F02C 7/055 (2013.01); **B64D 2033/022**
(2013.01); **F05D 2250/131** (2013.01); **Y02E**
50/672 (2013.01)

(58) **Field of Classification Search**
CPC **F02C 7/00**; **F02C 7/055**; **B64D 33/02**;
B64D 2033/022; **F05D 2250/131**; **Y02E**
50/672

See application file for complete search history.

(56) **References Cited**

U.S. PATENT DOCUMENTS

2,704,136 A * 3/1955 Rainbow B64D 33/02
55/306
2,944,631 A * 7/1960 Kerry B64D 33/02
55/306
3,121,545 A * 2/1964 Meletion B64D 33/02
244/53 B
3,196,598 A * 7/1965 Olson B64D 33/02
415/121.2
3,426,981 A * 2/1969 Allcock B64D 15/00
137/15.1
3,485,252 A * 12/1969 Brown F02C 7/042
137/15.1
3,568,694 A * 3/1971 Johnson B64D 33/02
137/15.1
3,905,566 A * 9/1975 Anderson B64D 33/02
244/53 B

(Continued)

Primary Examiner — Ehud Gartenberg

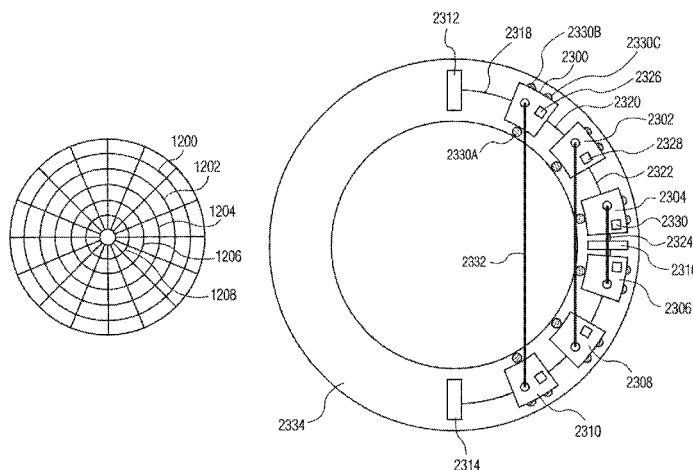
Assistant Examiner — Karthik Subramanian

(74) *Attorney, Agent, or Firm* — Karl F. Milde, Jr.; Eckert
Seamans Cherin & Mellott, LLC

(57) **ABSTRACT**

A retractable deflector to deflect birds and debris from an air intake duct of an aircraft jet engine. The duct has a forward opening for air receipt. The deflector includes a plurality of elongate members disposed on the duct in spaced relation to each other, each member having two end segments and a central segment disposed between the two end segments; and a plurality of guiding members, each mounted for movement along the perimeter of the duct and coupled to one end segment of an elongate member. The central segment of each elongate member extends between a pair of guiding members such that the elongate member is movable by a respective pair of guiding members between a retracted position and a deployed position in front of the duct. When in the deployed position, the central segments are situated to impede the ingress of debris into the duct.

23 Claims, 46 Drawing Sheets



(56)

References Cited

U.S. PATENT DOCUMENTS

4,004,760	A *	1/1977	Ando	B64D 33/02 244/53 B
4,070,827	A *	1/1978	Vanfleet	F02C 7/05 415/121.2
4,137,535	A *	1/1979	Rupprecht	H01Q 1/1235 343/902
4,354,346	A *	10/1982	Wooding	B64D 33/02 55/306
5,102,375	A *	4/1992	Featherstone	B66F 3/06 227/99
5,123,240	A *	6/1992	Frost	F02K 7/075 244/53 B
5,139,464	A *	8/1992	Lehnert	E04H 12/185 474/155
5,385,612	A *	1/1995	Li	B60R 1/0602 134/113
5,411,224	A *	5/1995	Dearman	B64D 33/02 244/53 B
5,779,169	A *	7/1998	Sloan	B64C 15/14 239/265.25
D433,029	S *	10/2000	Eidson	D12/345
6,872,232	B1 *	3/2005	Pavlatos	B01D 45/12 55/306
6,883,751	B2 *	4/2005	Koncsek	B64D 33/02 244/53 B
6,910,327	B2 *	6/2005	Sakurai	B64D 33/02 137/15.1
6,945,494	B2 *	9/2005	Bagnall	B64D 33/02 244/53 B
6,994,738	B2 *	2/2006	Taddey	B01D 45/04 55/306
D631,420	S *	1/2011	Locklear	D12/345
7,871,455	B1 *	1/2011	Sands	B64D 33/02 244/53 B
8,052,083	B1 *	11/2011	Moran	B64D 33/02 244/53 B
8,117,820	B1 *	2/2012	Briscoe	F02C 7/05 137/15.1
2004/0144097	A1 *	7/2004	Grimlund	F02C 7/042 60/772
2010/0270427	A1 *	10/2010	Barrientos	B64D 33/02 244/1 R
2011/0000184	A1 *	1/2011	Baugh	B64D 33/02 60/39.092
2011/0011055	A1 *	1/2011	Troy	F02C 7/055 60/39.092

* cited by examiner

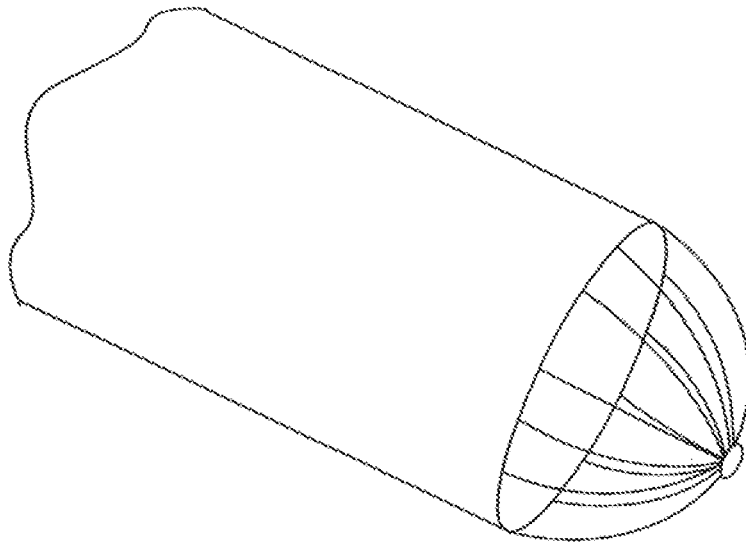


FIG. 1A

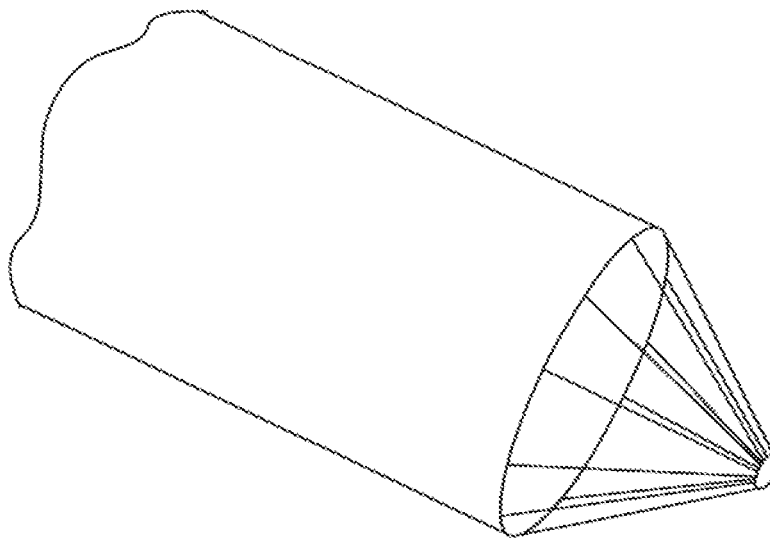


FIG. 1B

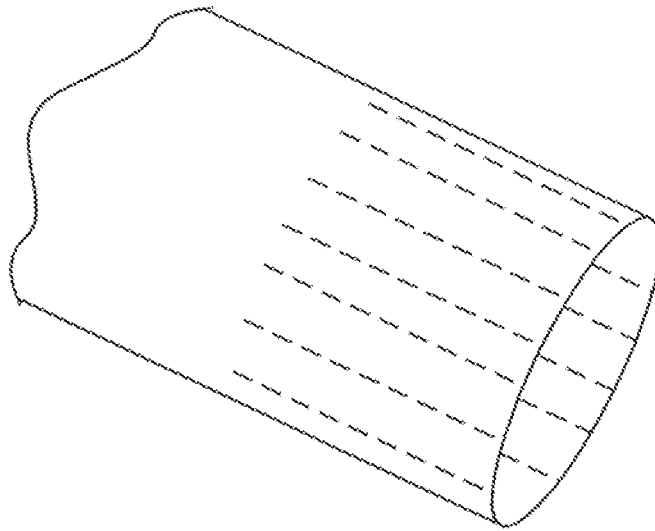


FIG. 1C

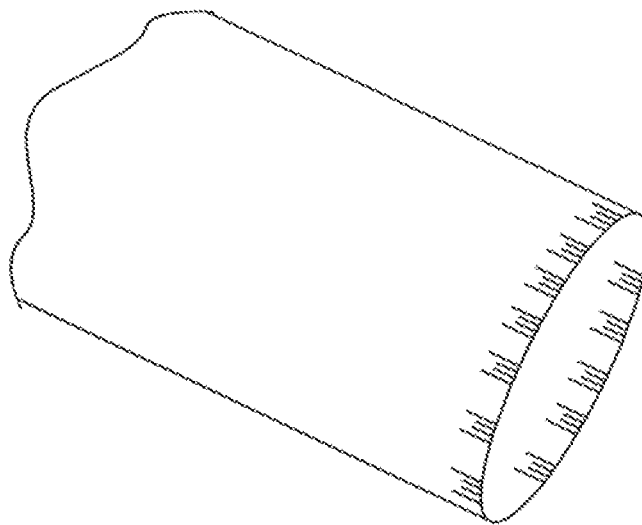


FIG. 1D

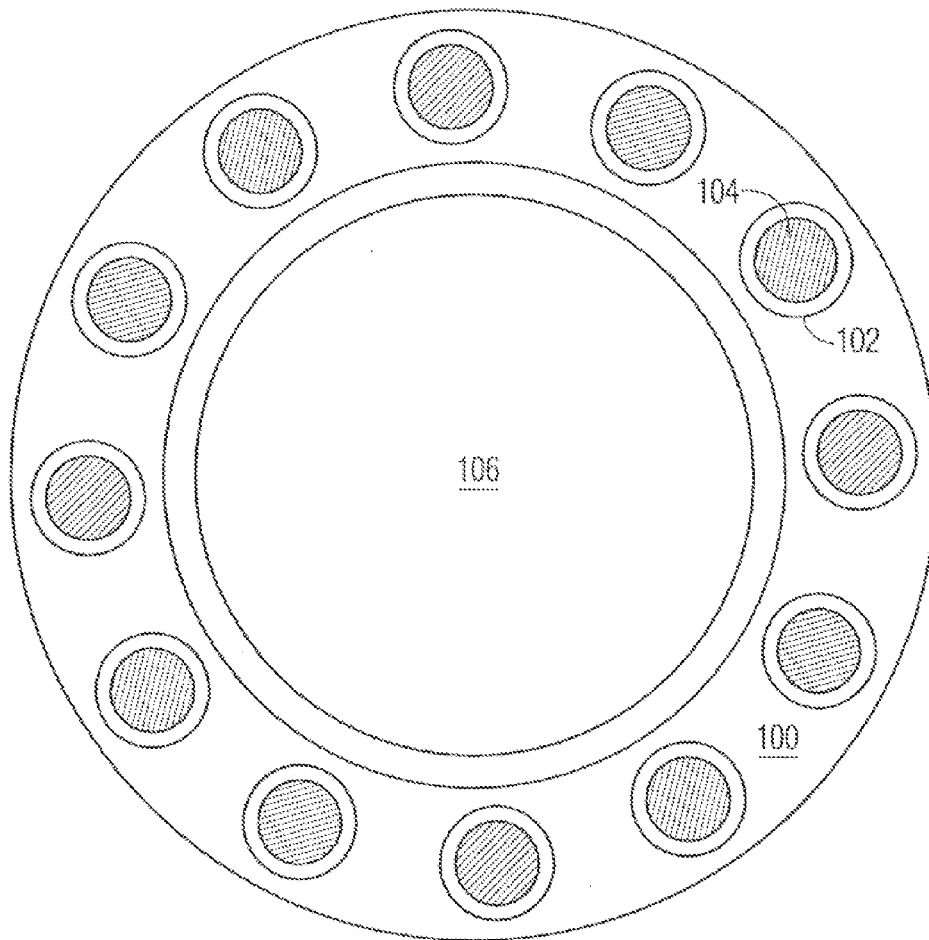


FIG. 1E

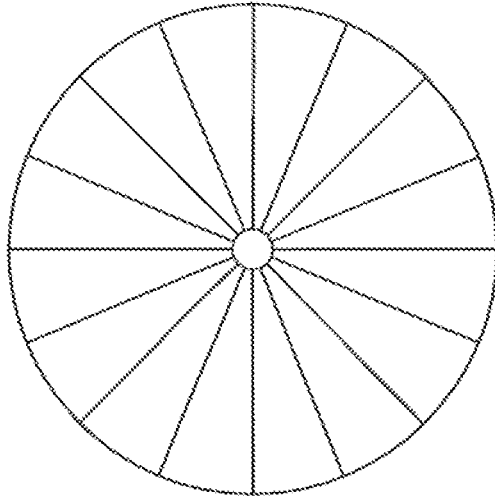


FIG. 2A

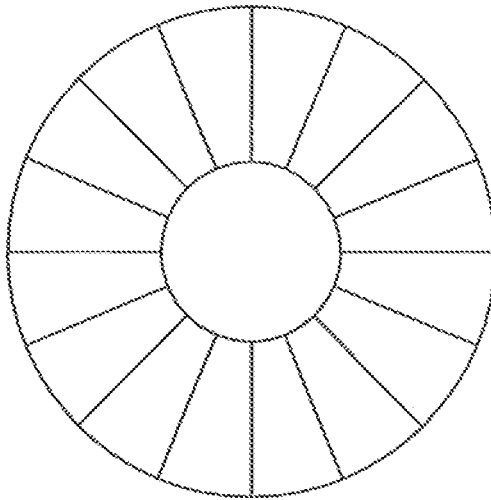


FIG. 2B

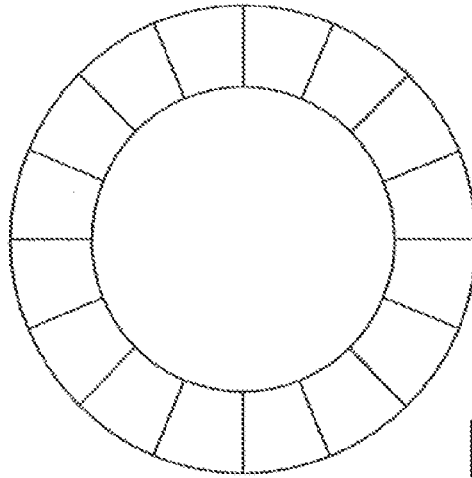


FIG. 2C

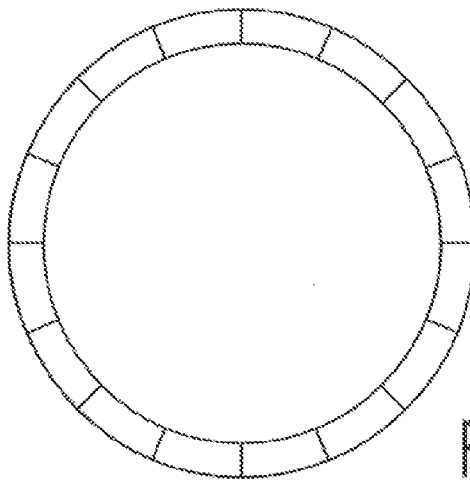


FIG. 2D

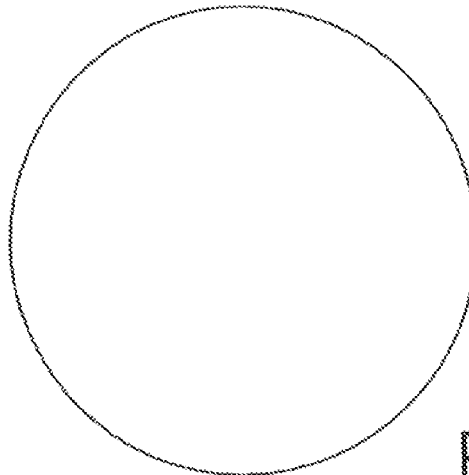


FIG. 2E

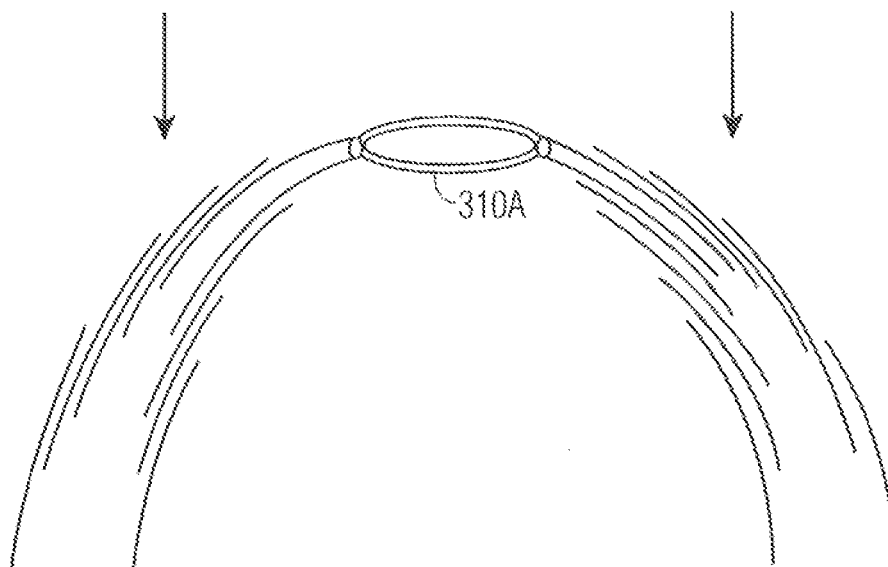


FIG. 3A

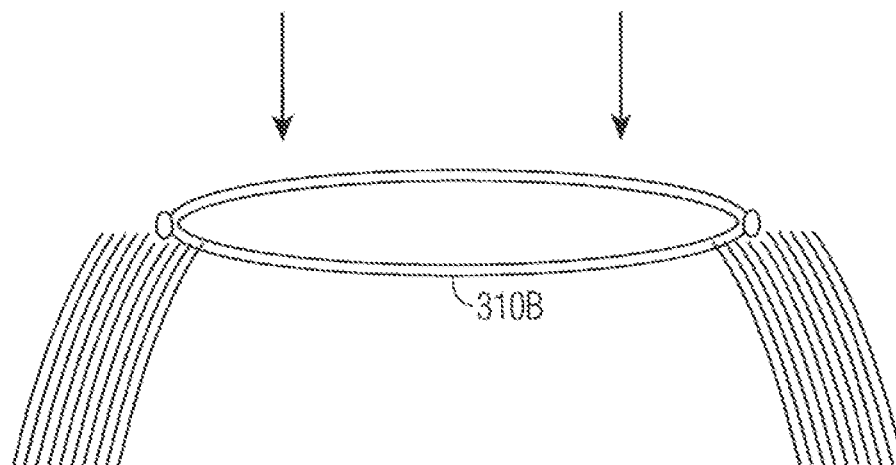


FIG. 3B

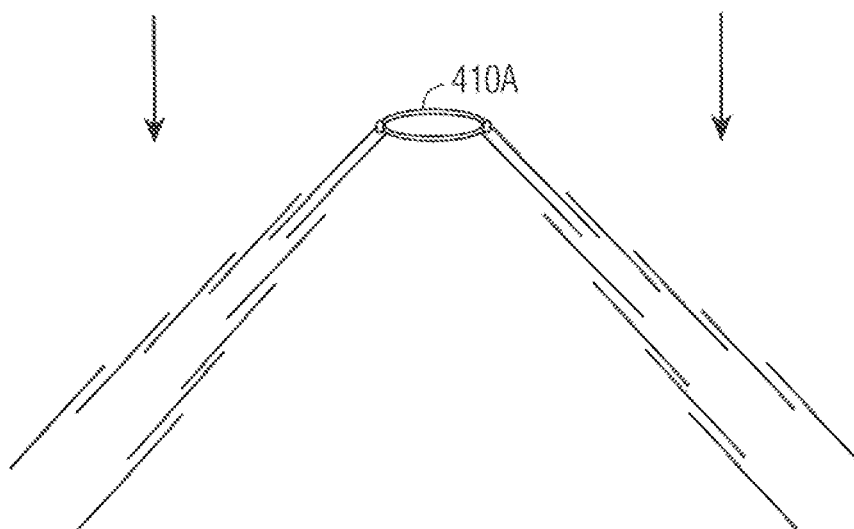


FIG. 4A

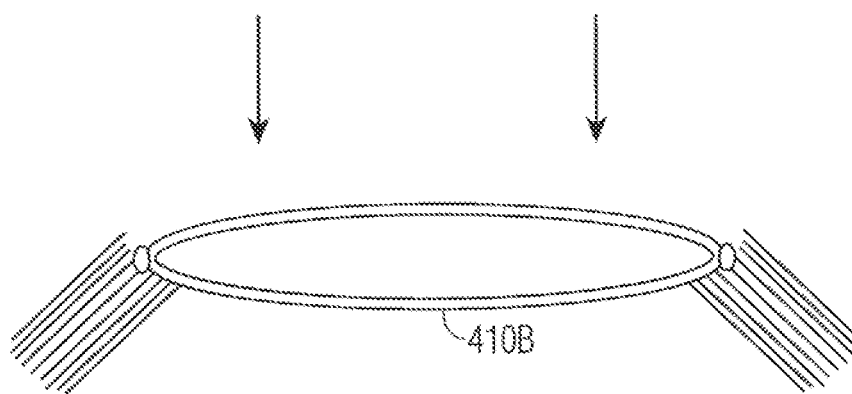


FIG. 4B

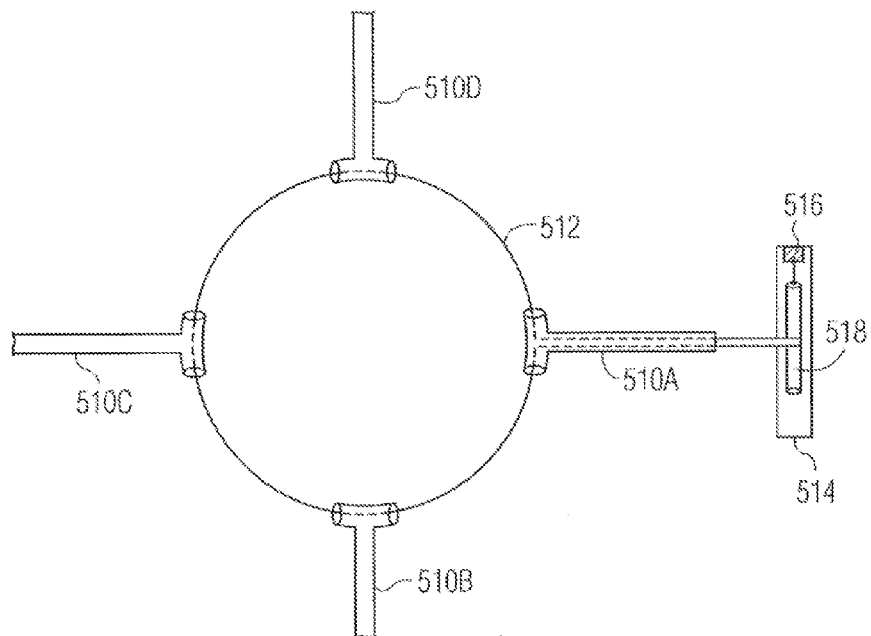


FIG. 5A

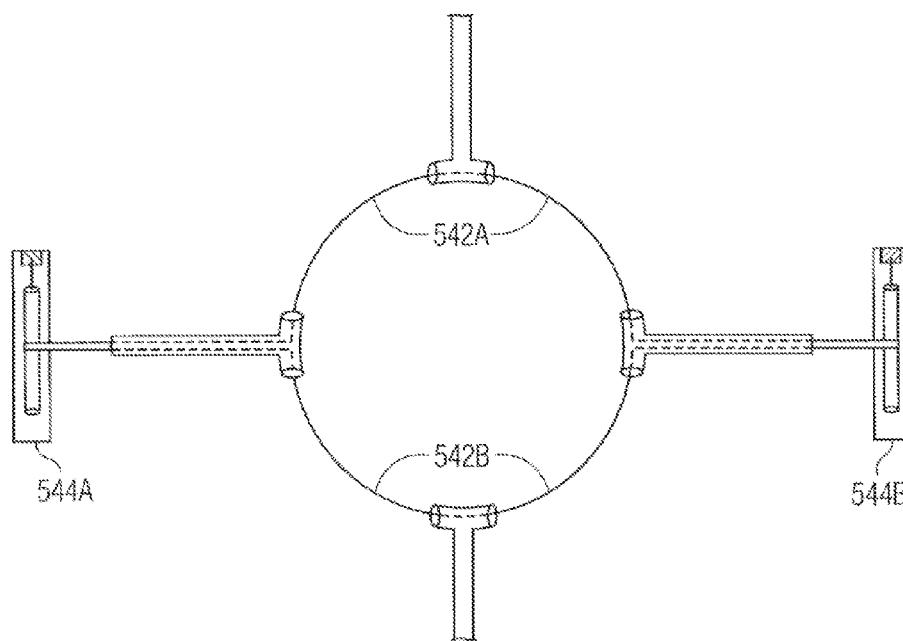


FIG. 5B

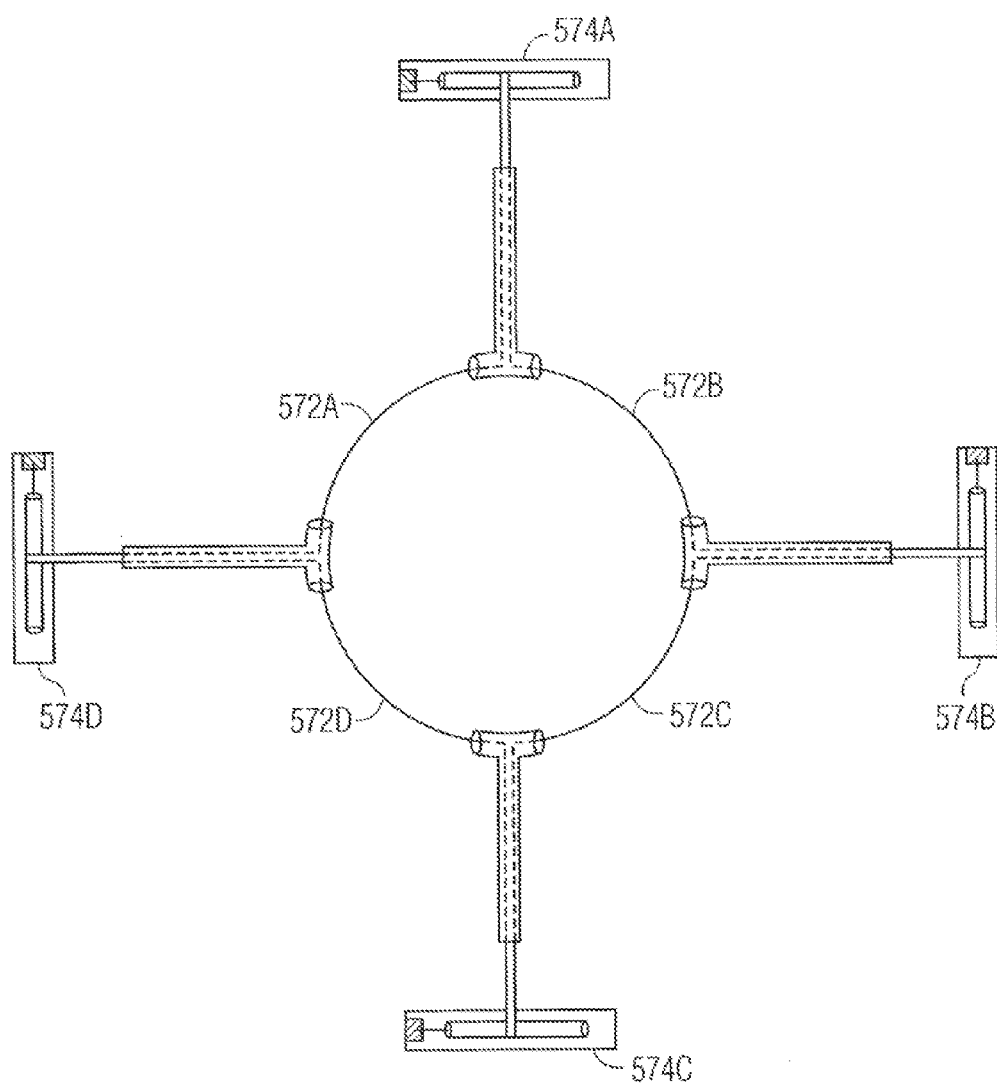


FIG. 5C

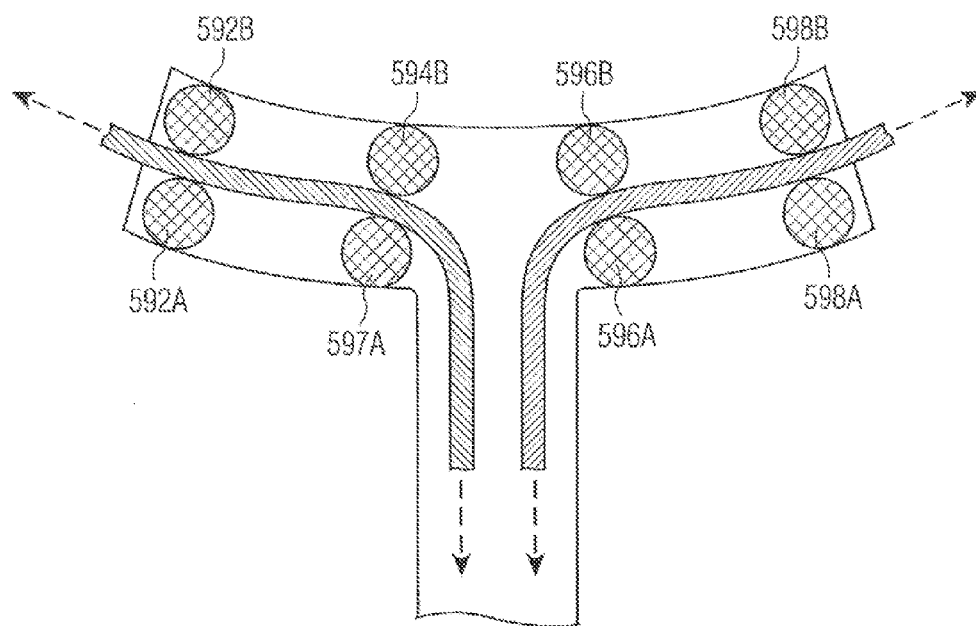


FIG. 5D



FIG. 6A

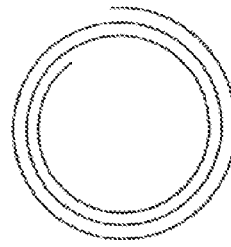


FIG. 6B

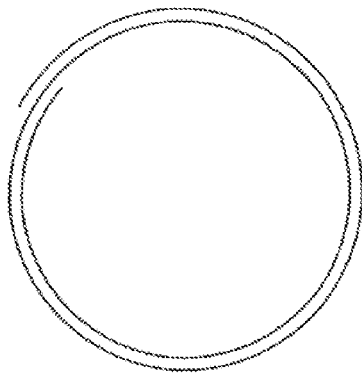


FIG. 6C

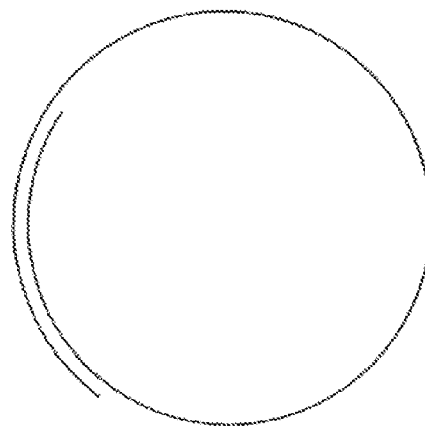


FIG. 6D

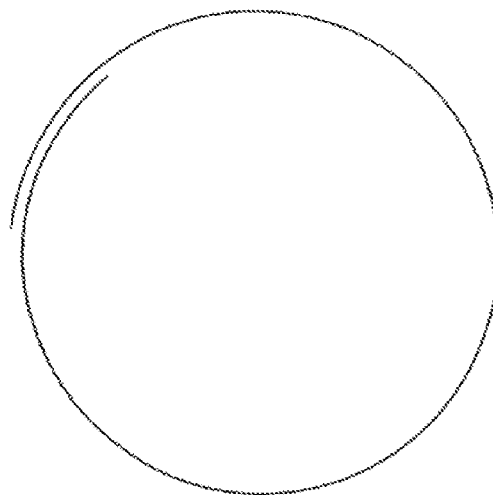


FIG. 6E

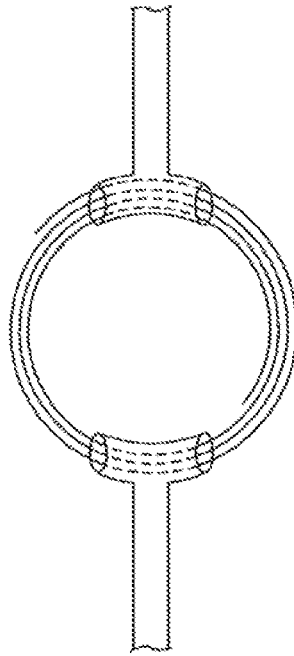


FIG. 7A

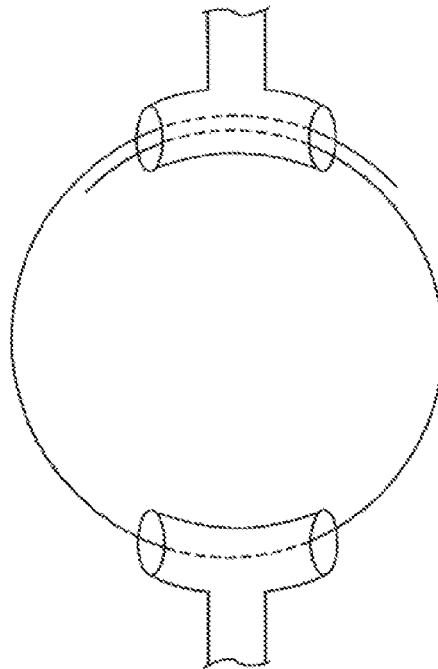


FIG. 7B

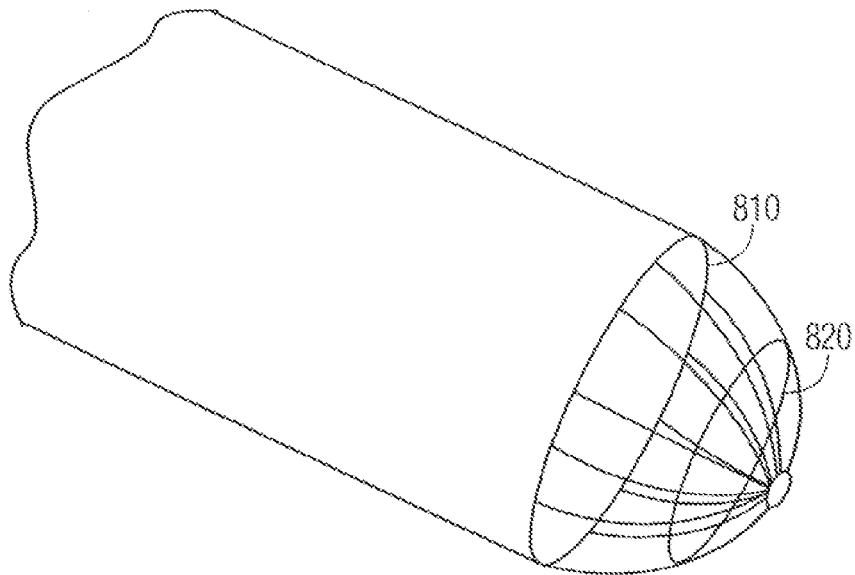


FIG. 8A

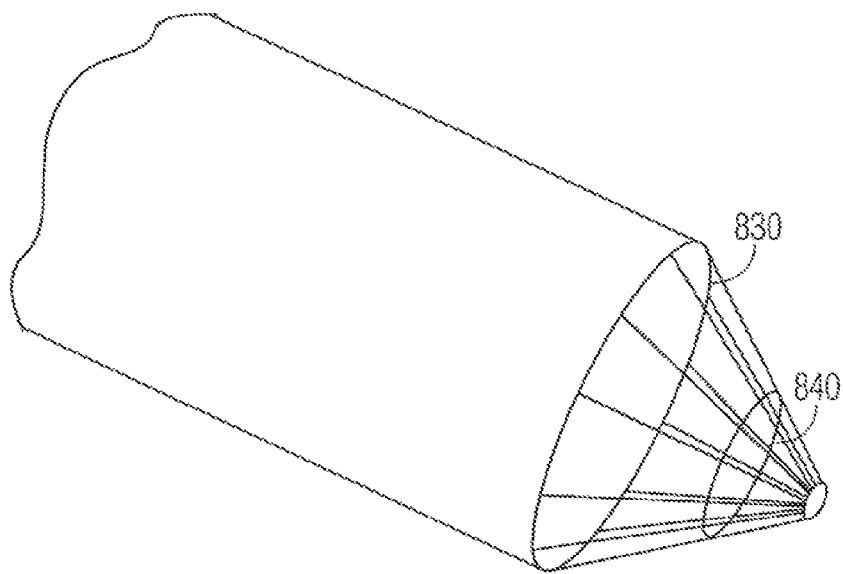


FIG. 8B

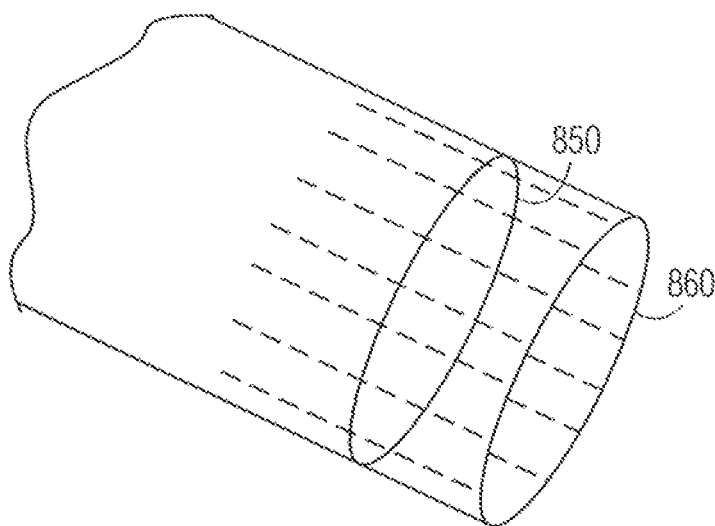


FIG. 8C

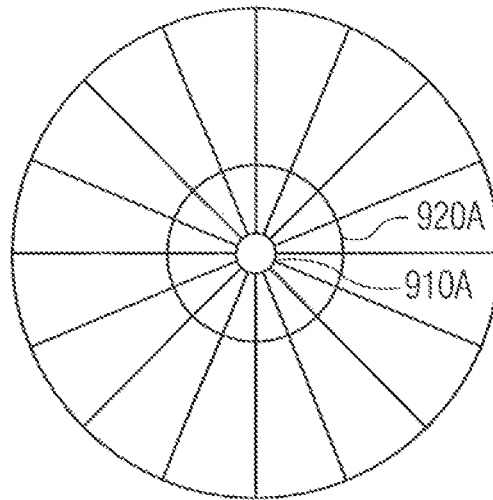


FIG. 9A

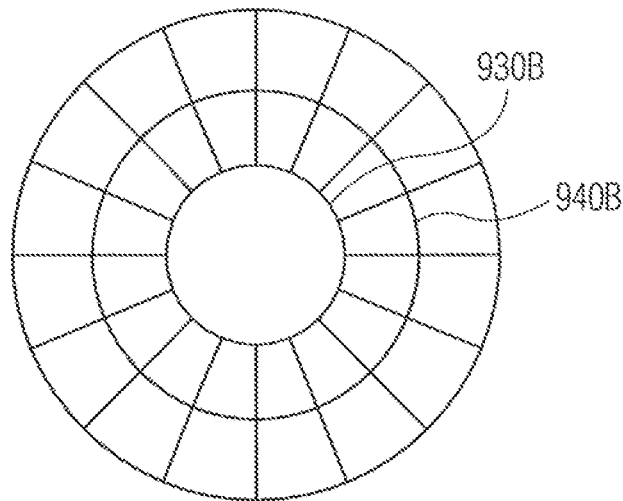


FIG. 9B

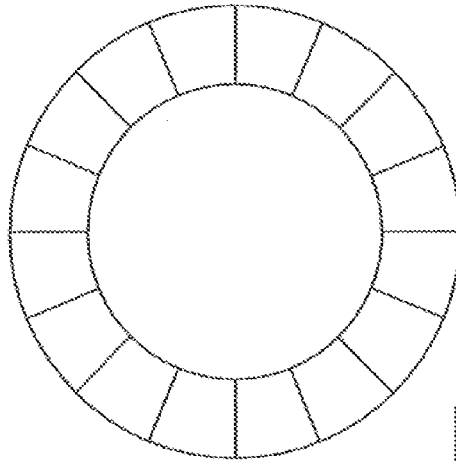


FIG. 9C

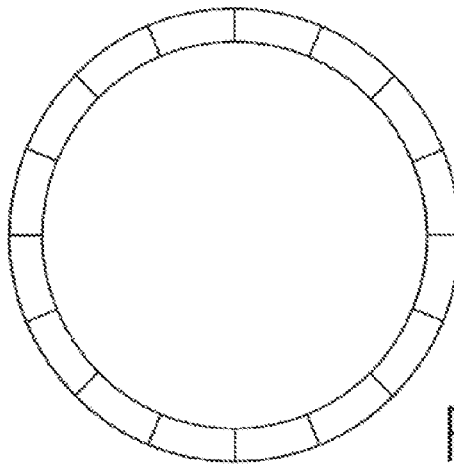


FIG. 9D

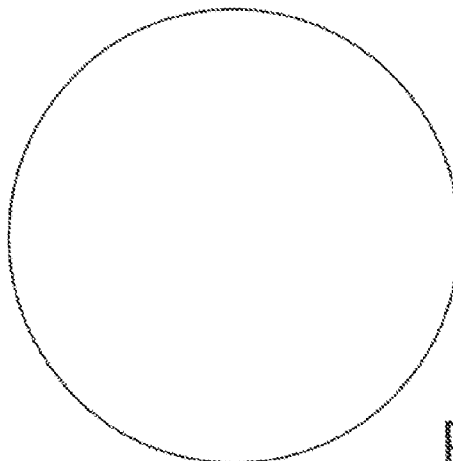


FIG. 9E

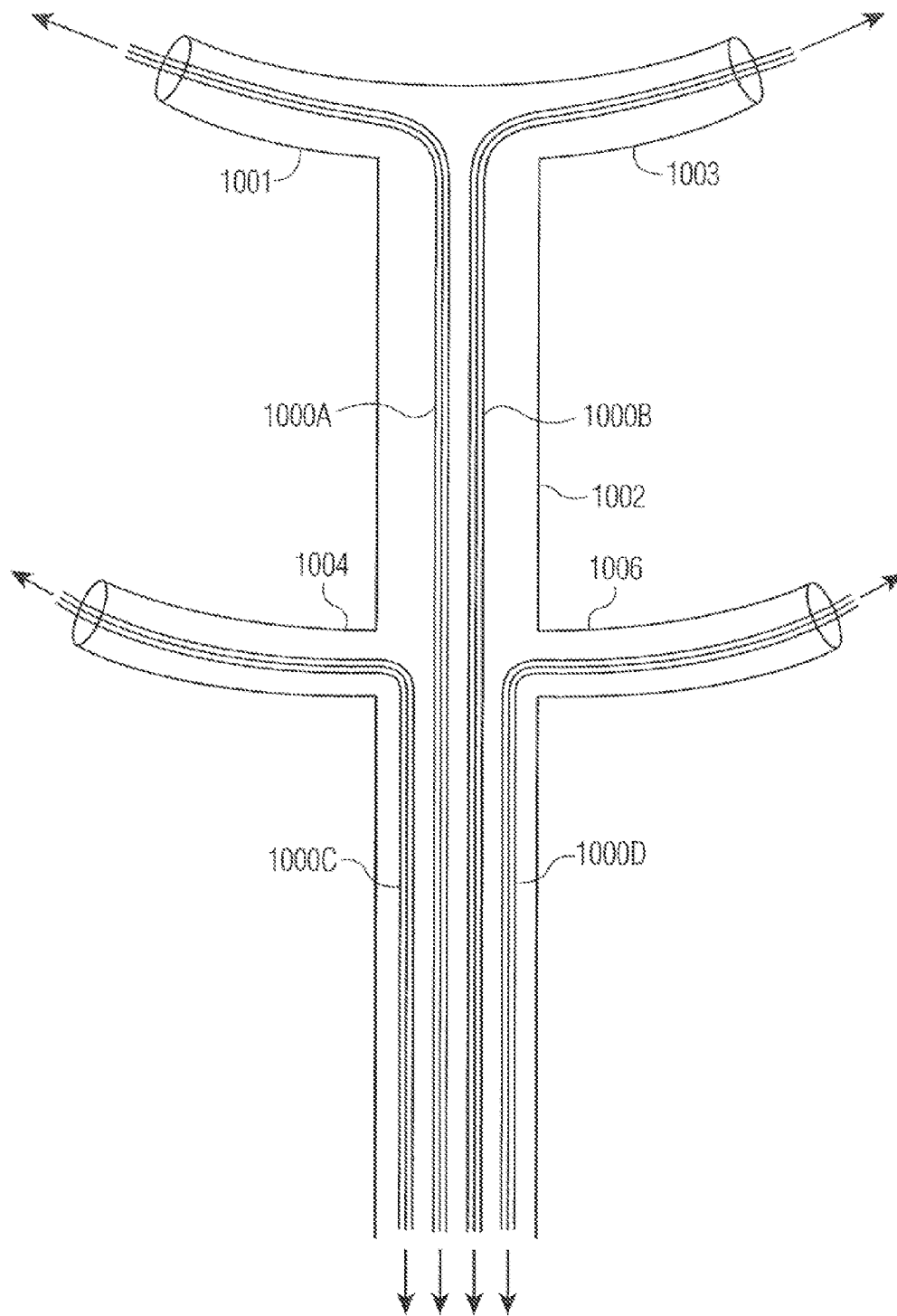


FIG. 10A

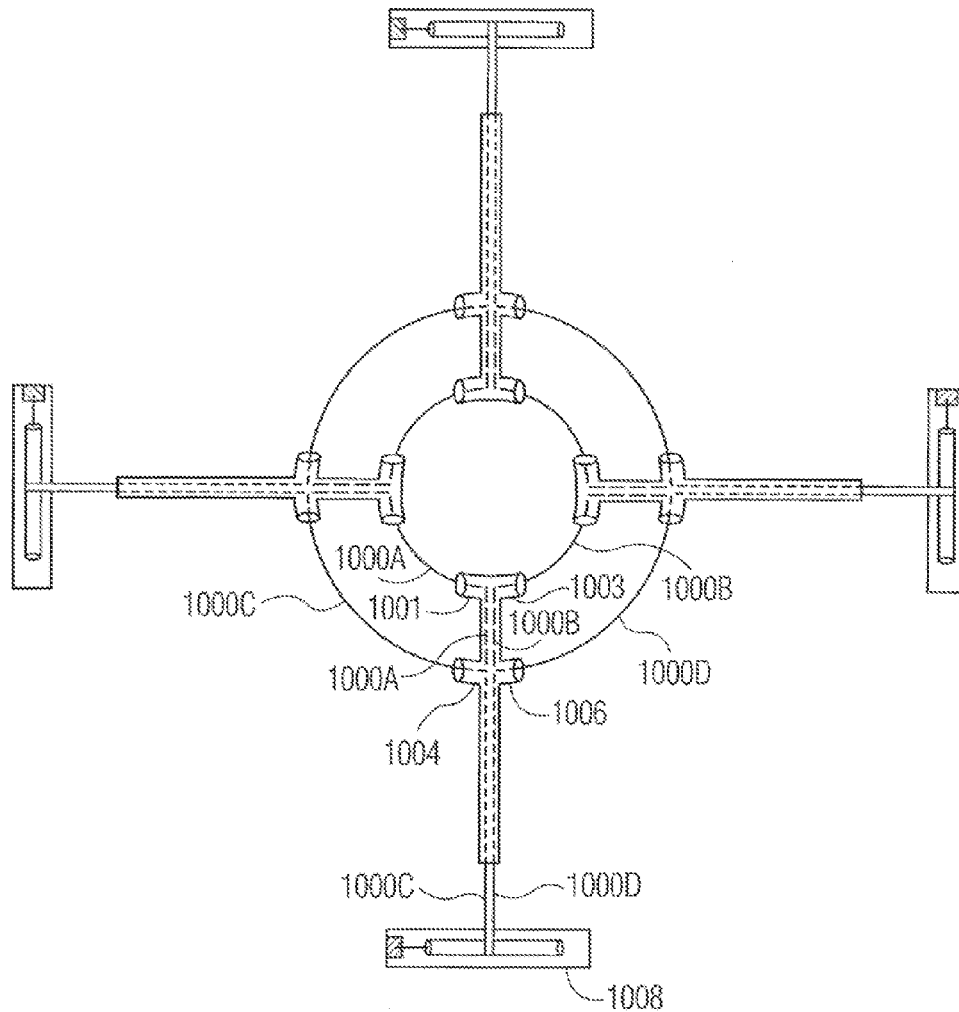


FIG. 10B

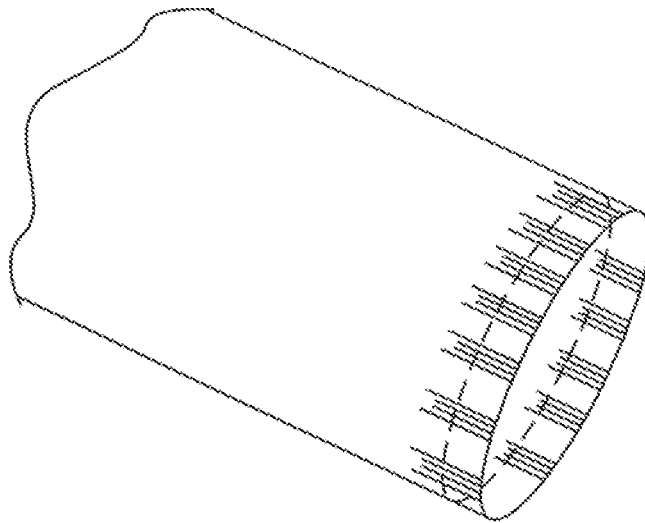


FIG. 10C

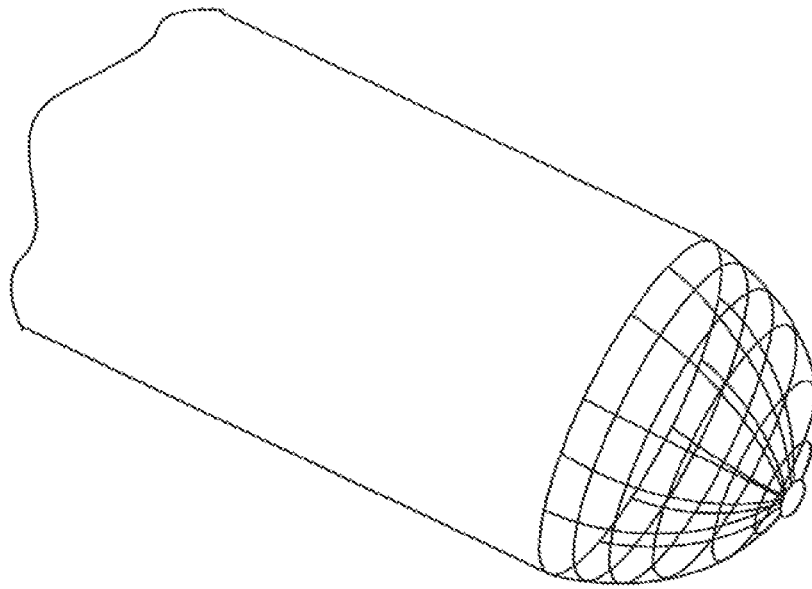


FIG. 11A

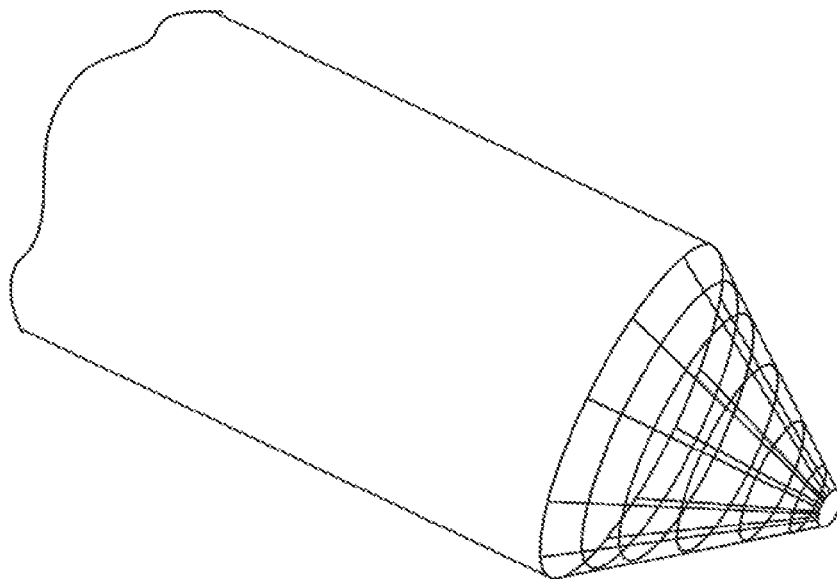


FIG. 11B

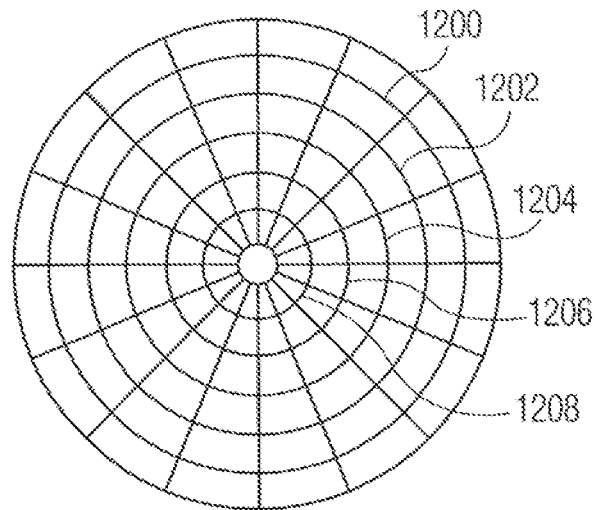


FIG. 12A

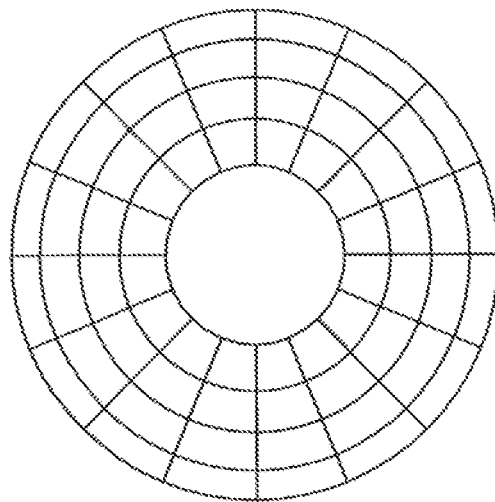


FIG. 12B

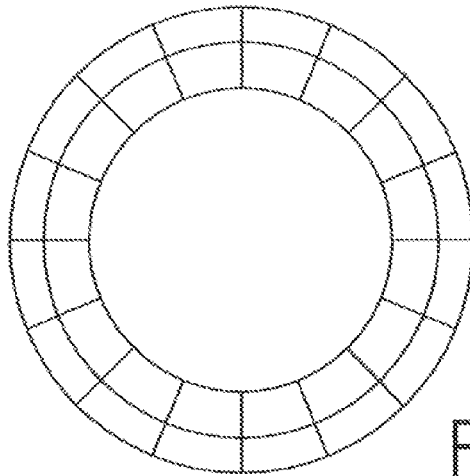


FIG. 12C

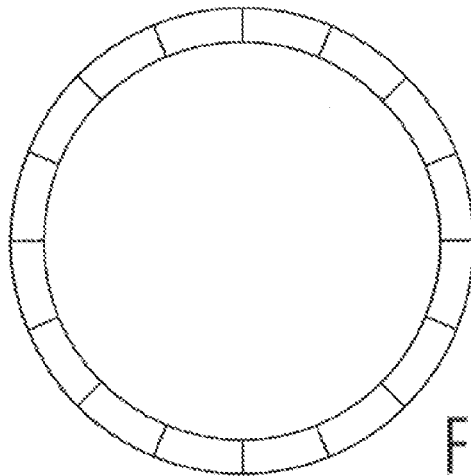


FIG. 12D

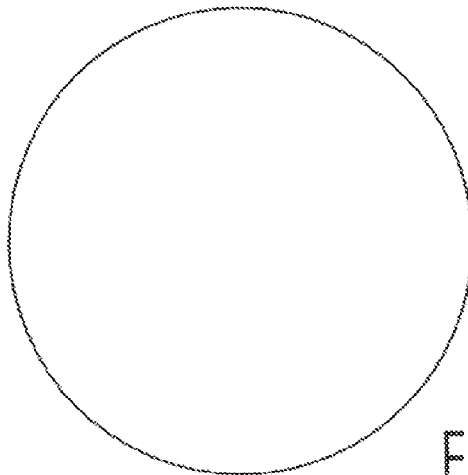


FIG. 12E

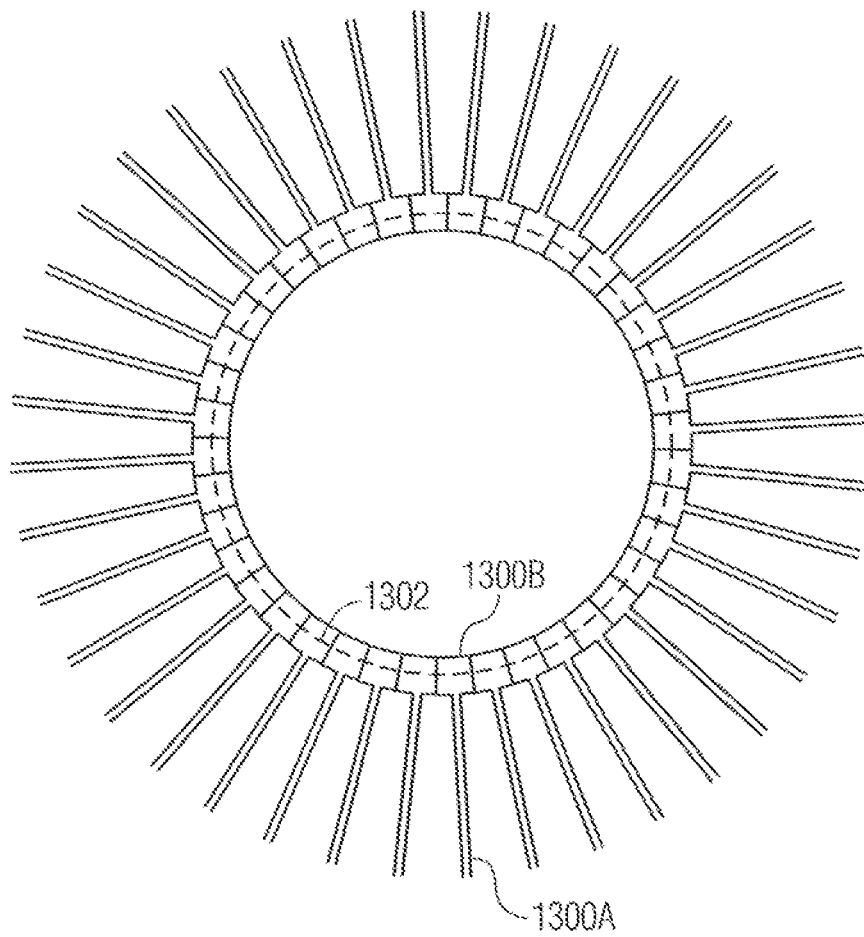


FIG. 13

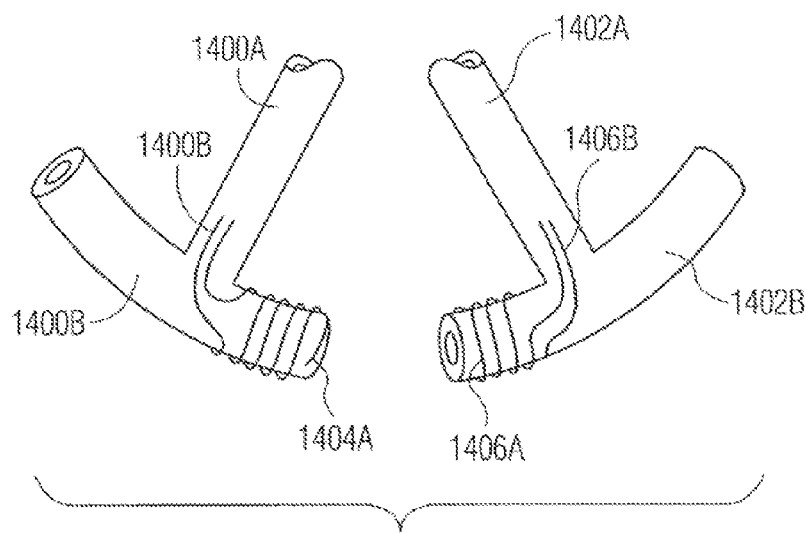


FIG. 14

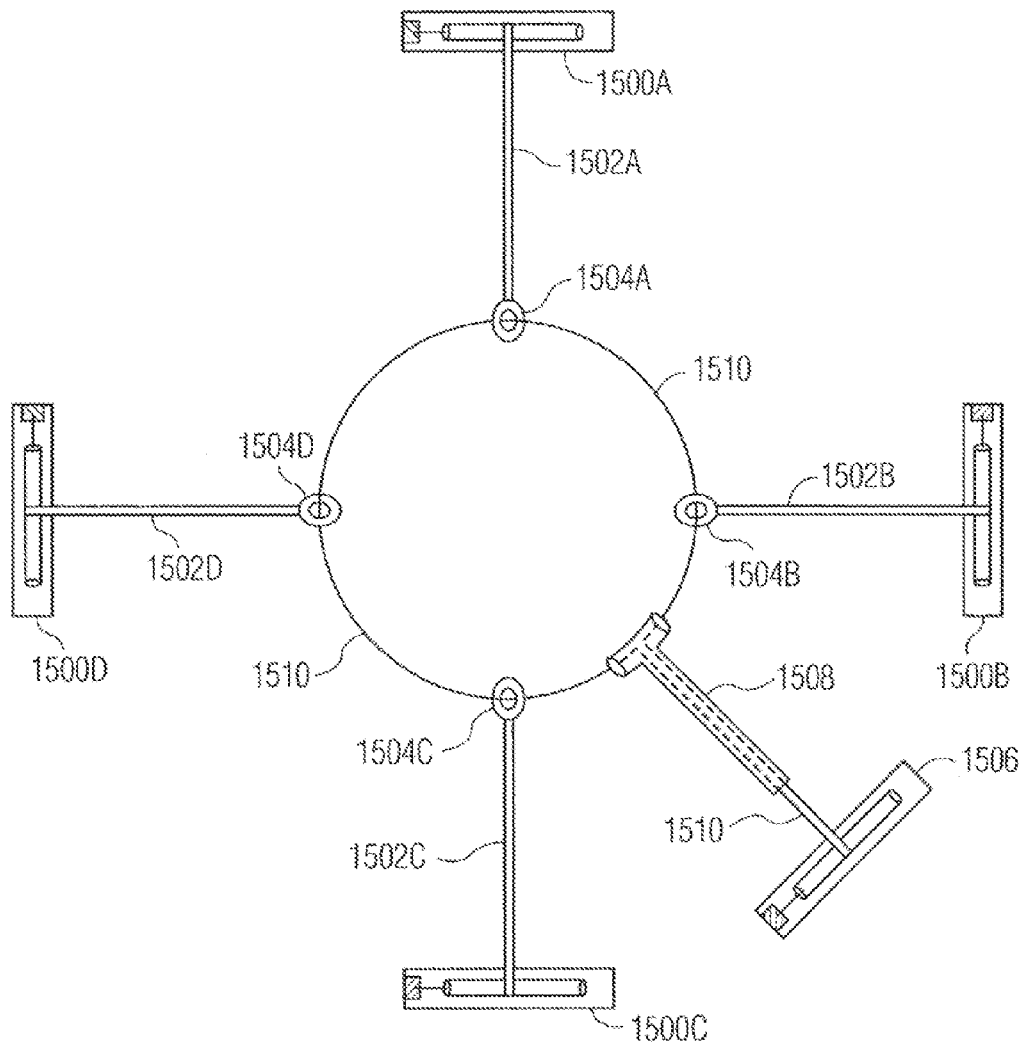


FIG. 15A

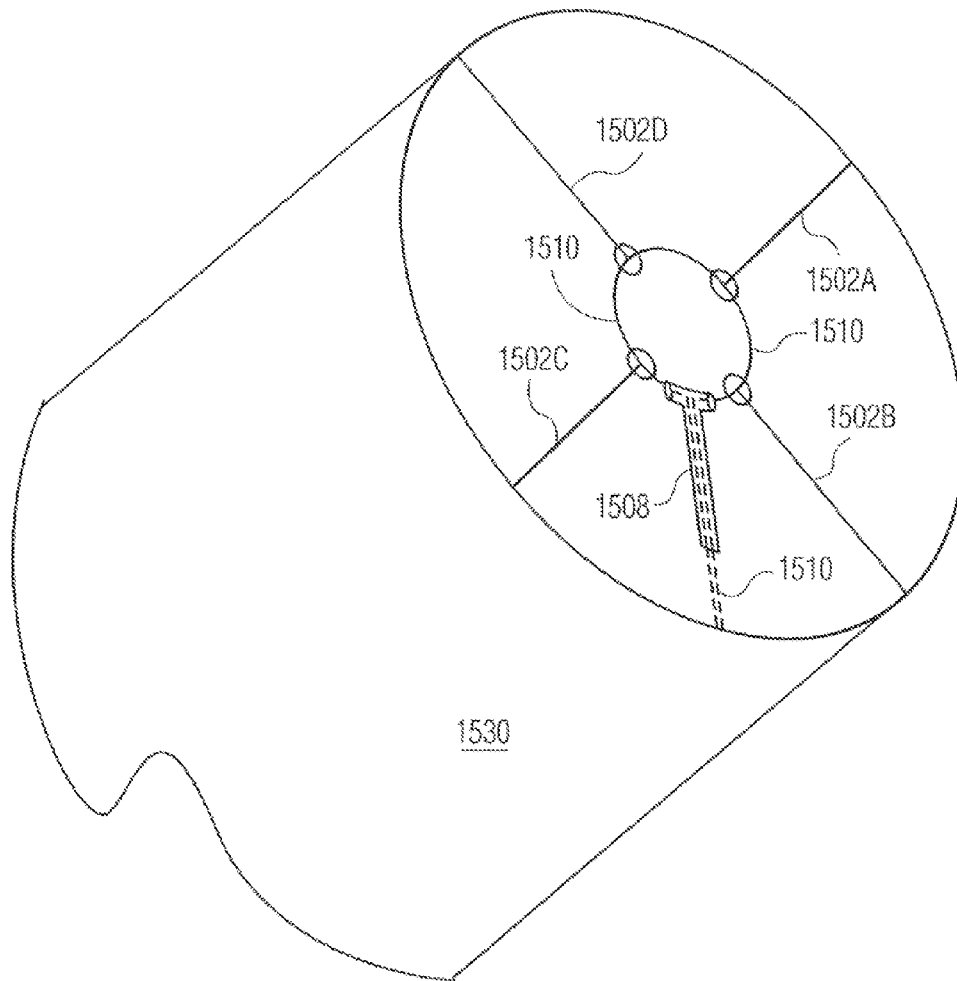


FIG. 15B

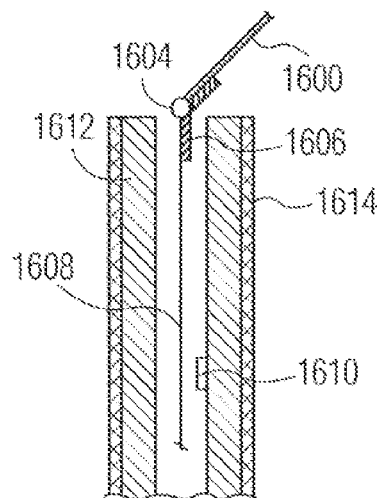


FIG. 16A

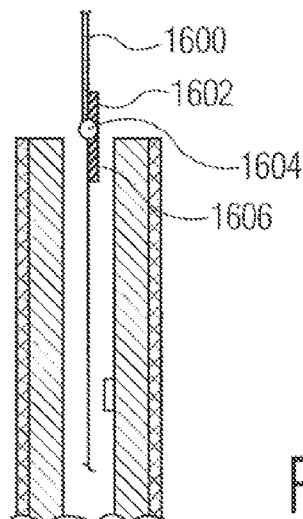


FIG. 16B

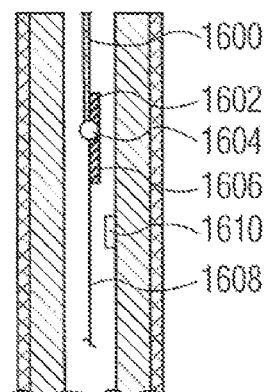


FIG. 16C

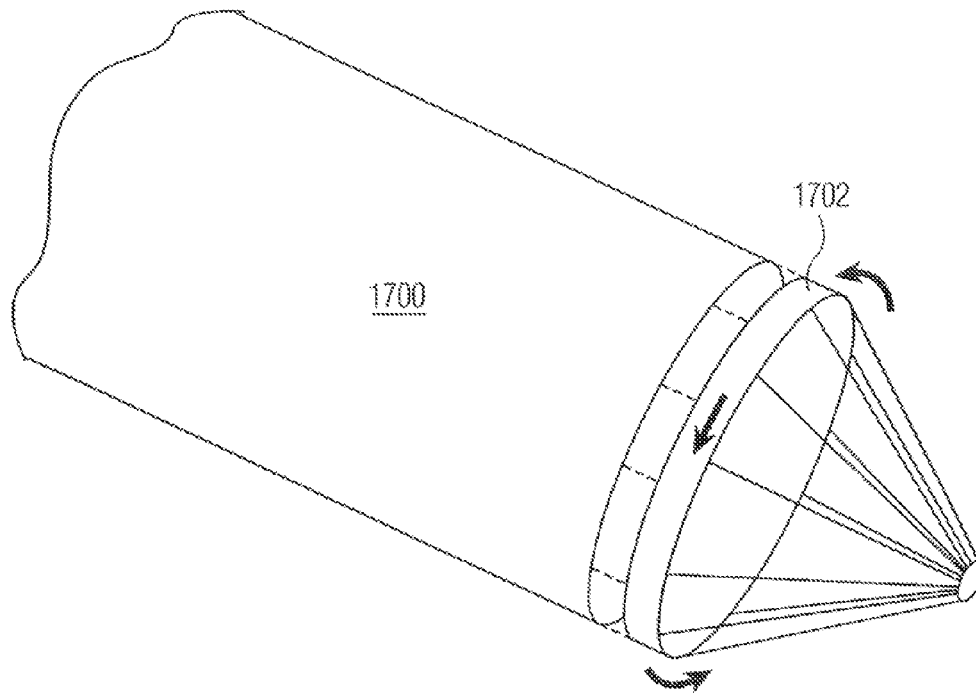


FIG. 17

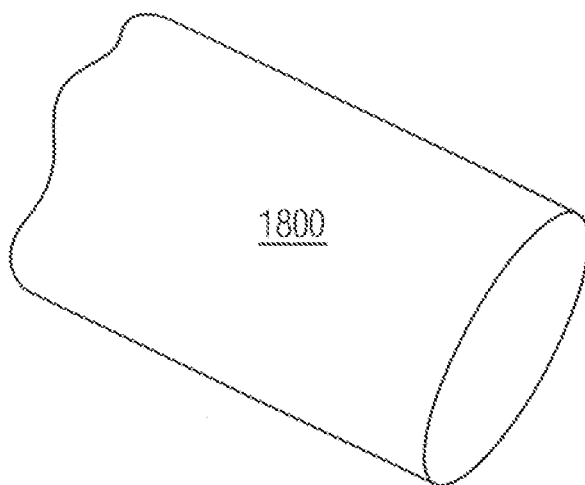


FIG. 18A

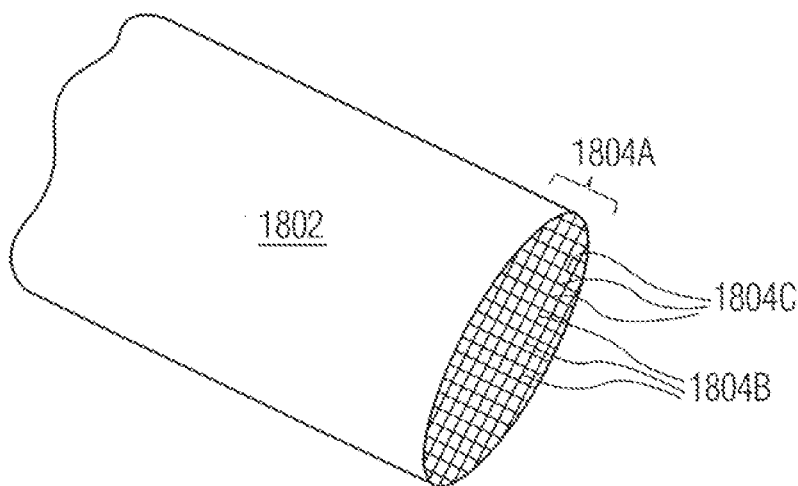


FIG. 18B

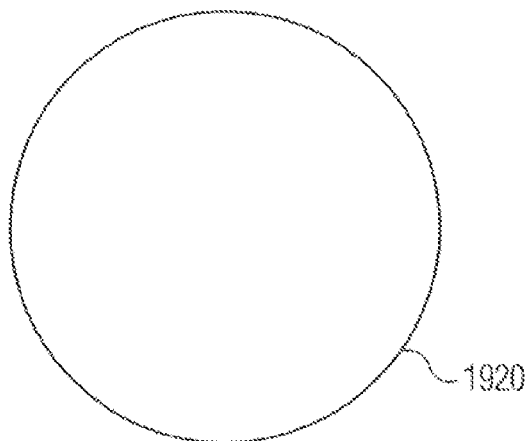


FIG. 19A

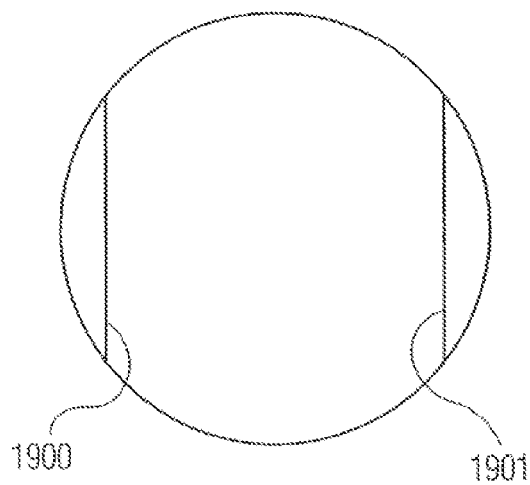


FIG. 19B

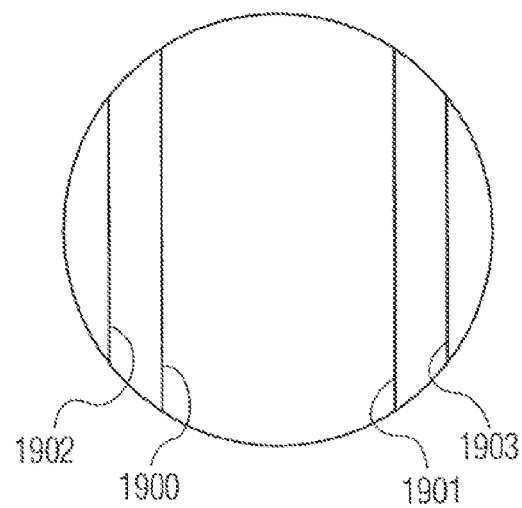


FIG. 19C

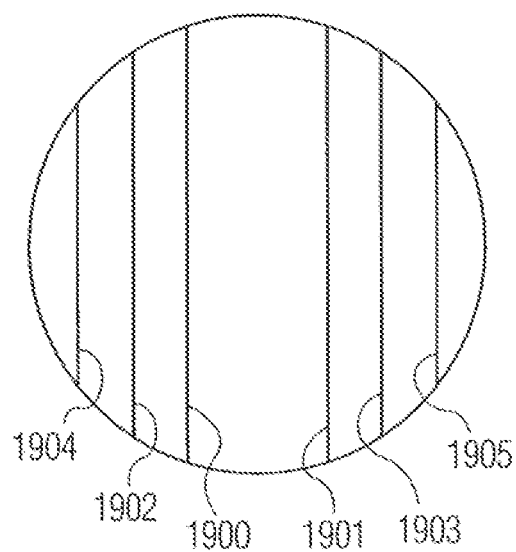


FIG. 19D

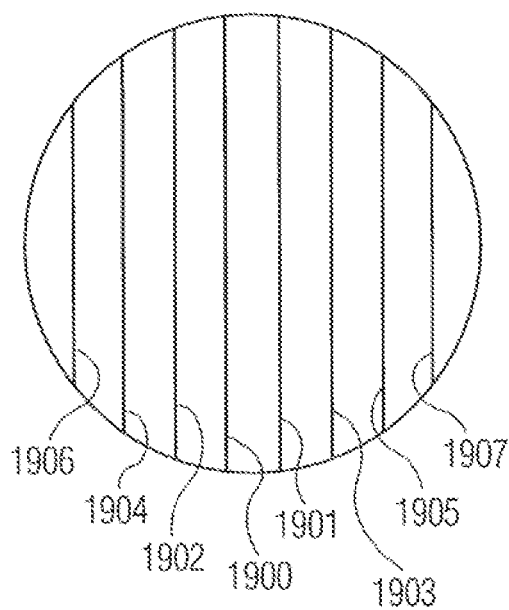


FIG. 19E

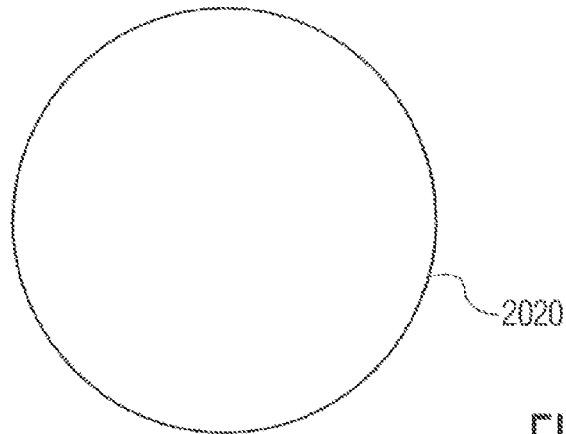


FIG. 20A

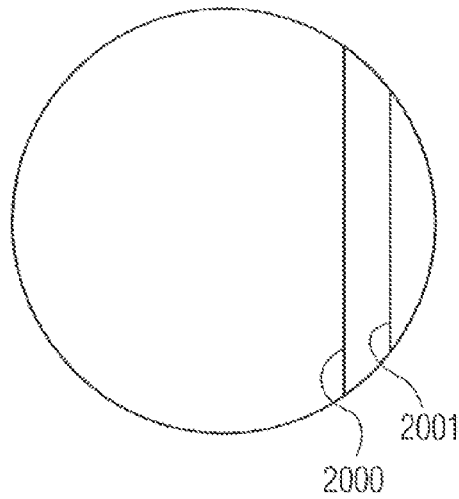


FIG. 20B

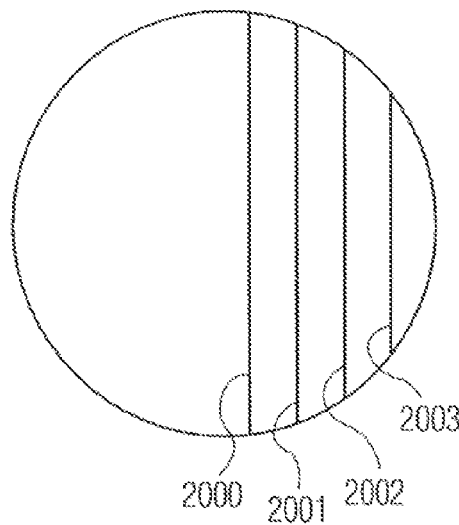


FIG. 20C

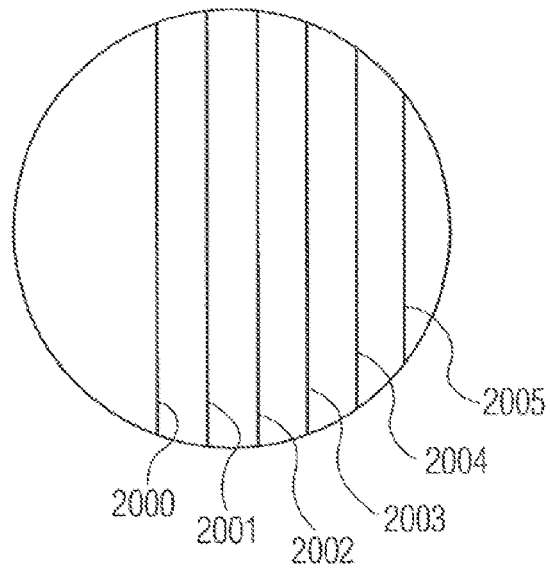


FIG. 20D

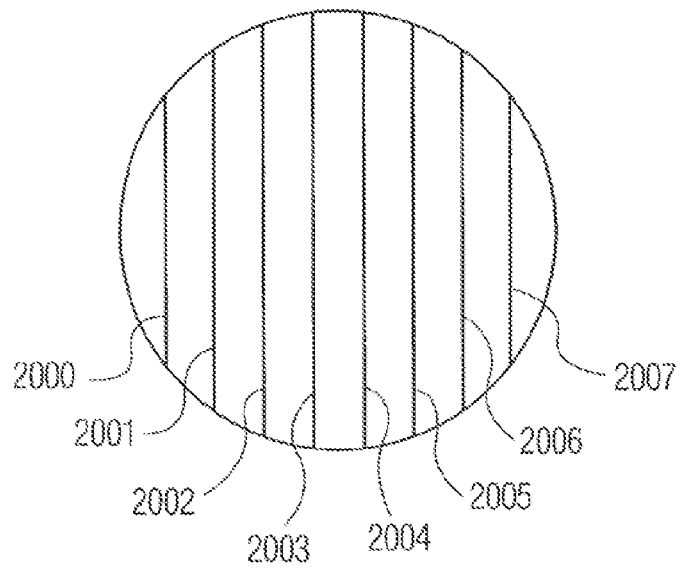


FIG. 20E

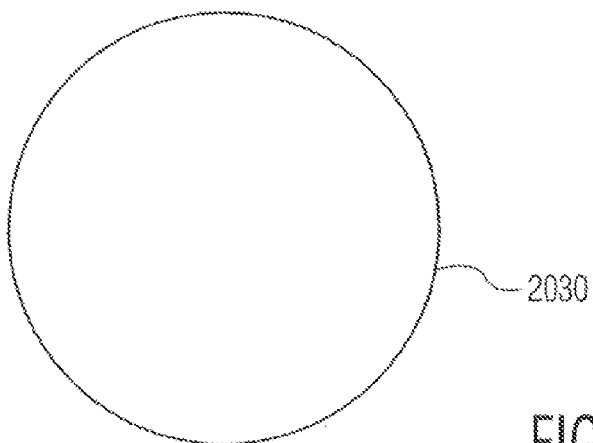


FIG. 20F

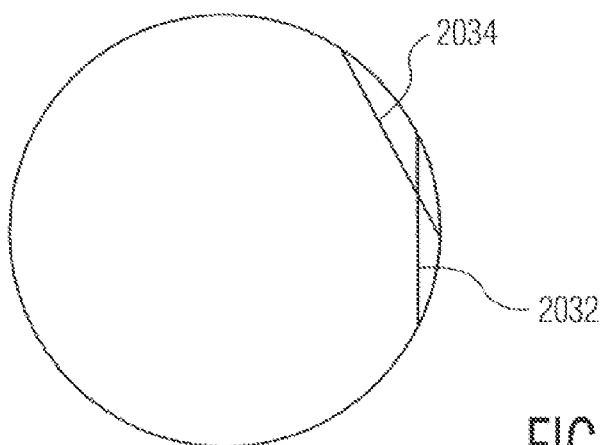


FIG. 20G

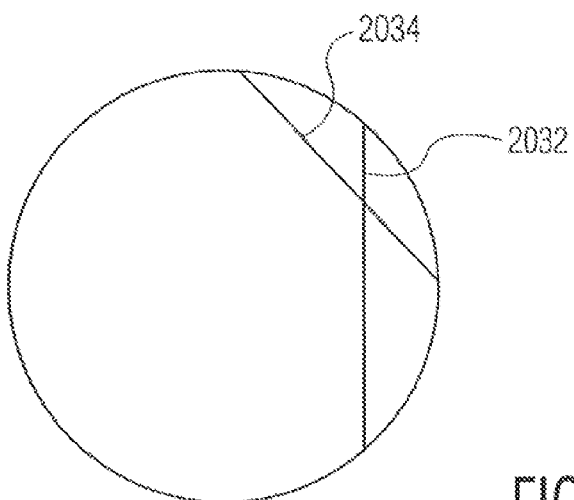


FIG. 20H

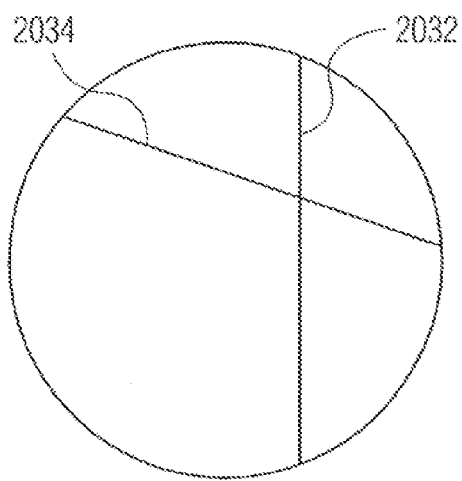


FIG. 20I

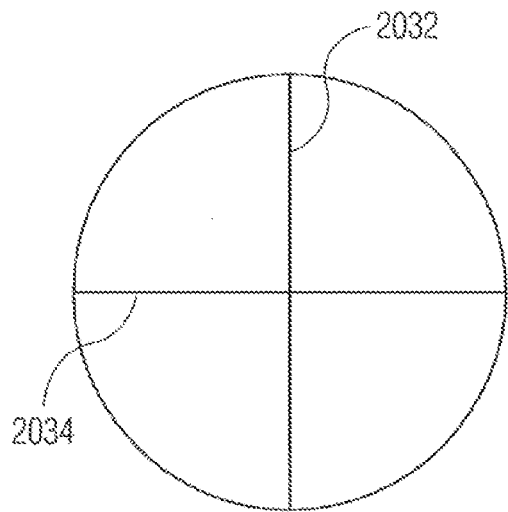


FIG. 20J

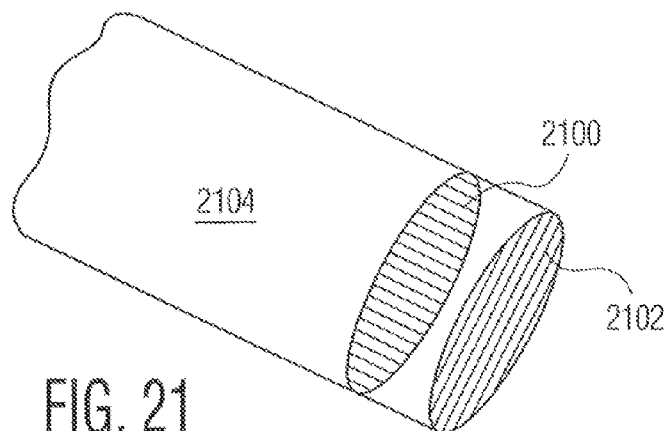


FIG. 21

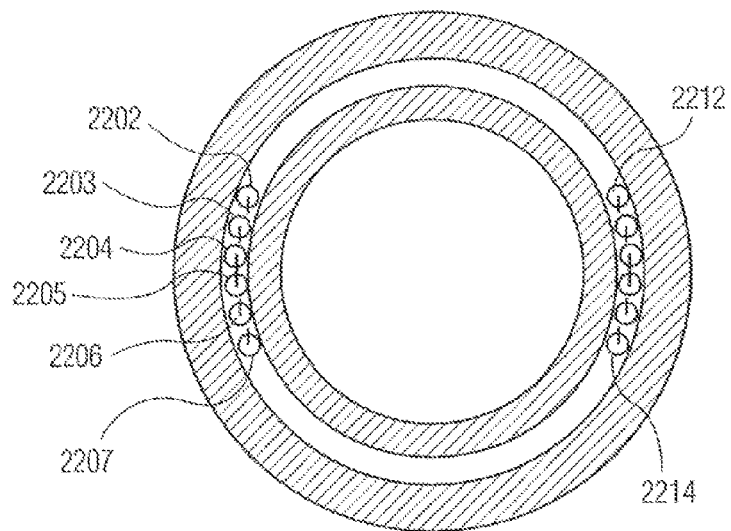


FIG. 22A

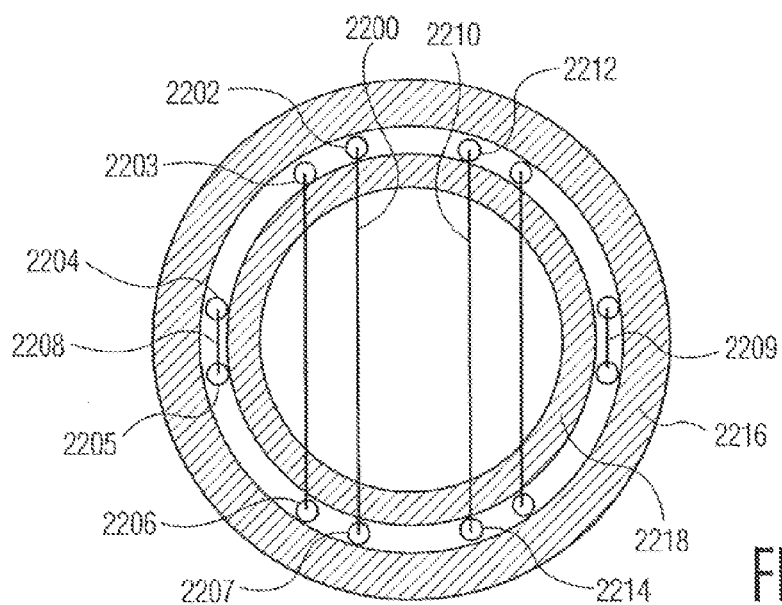


FIG. 22B

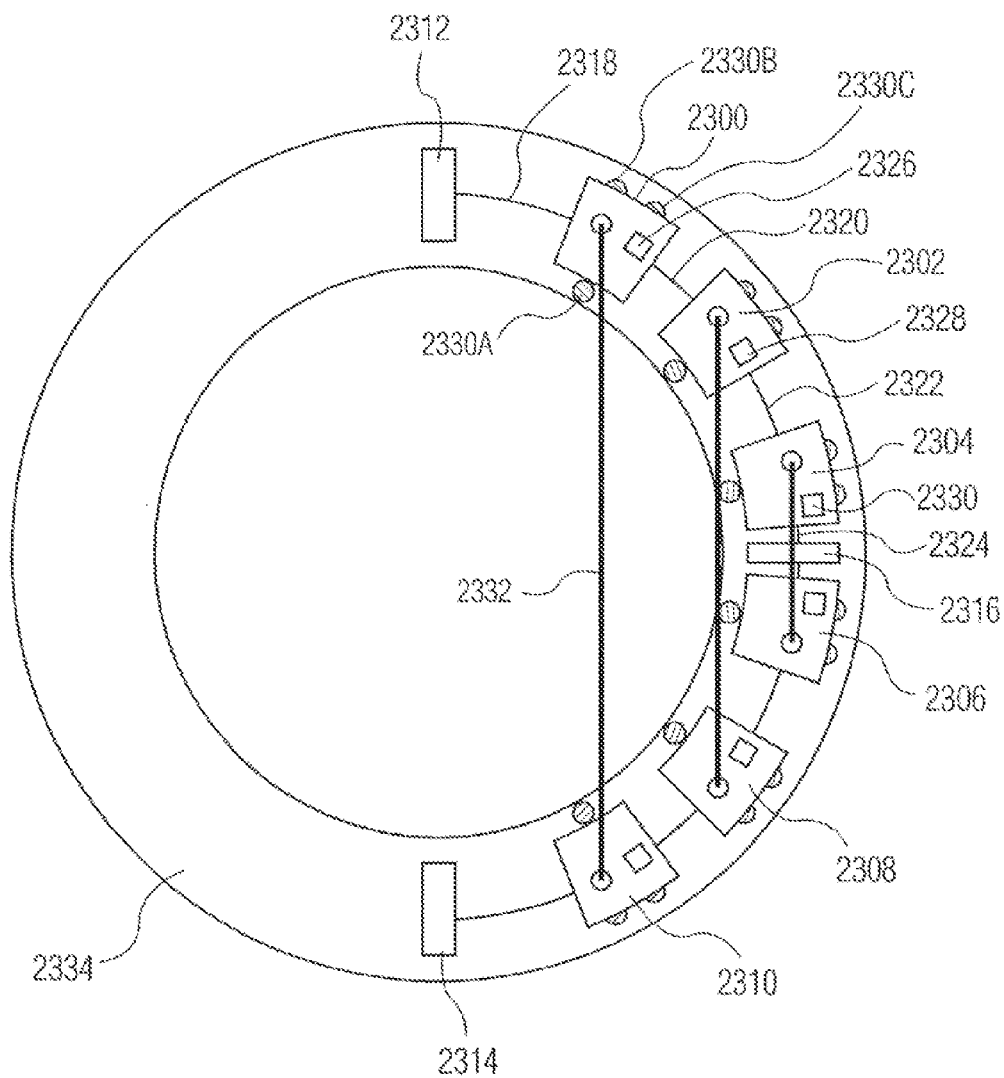


FIG. 23A

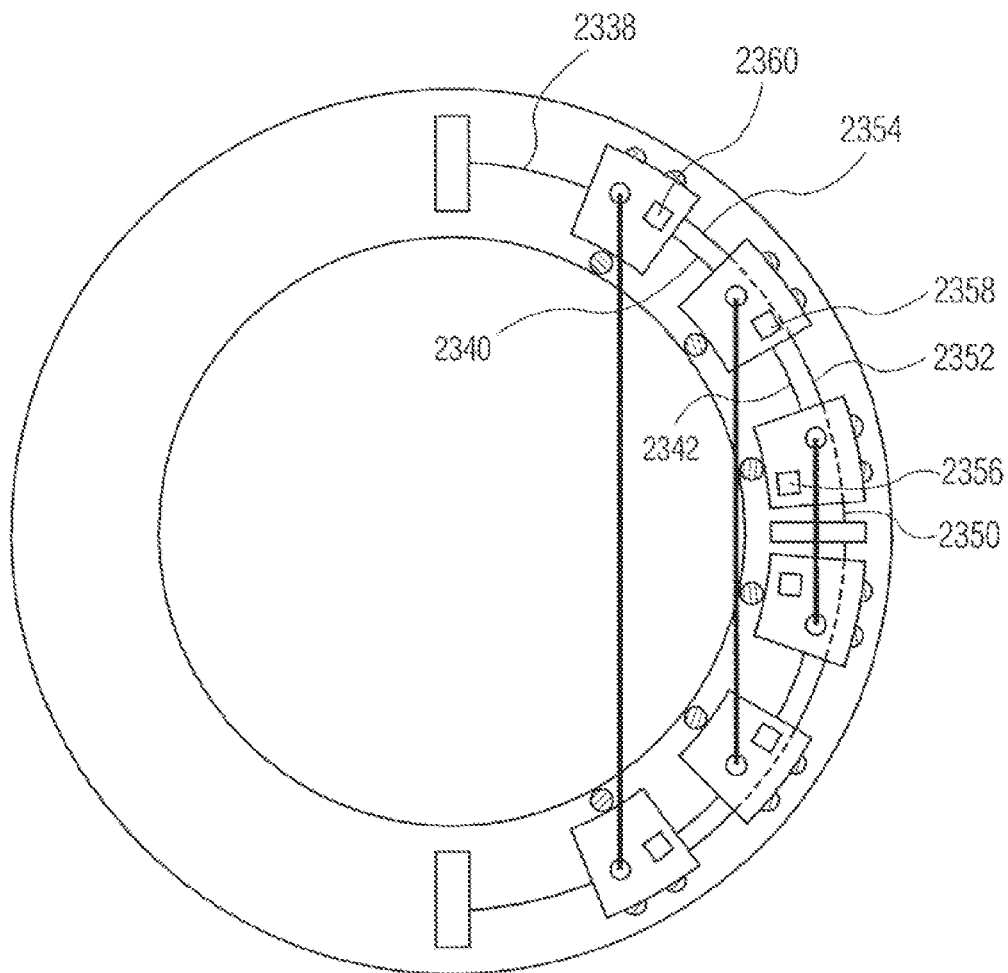


FIG. 23B

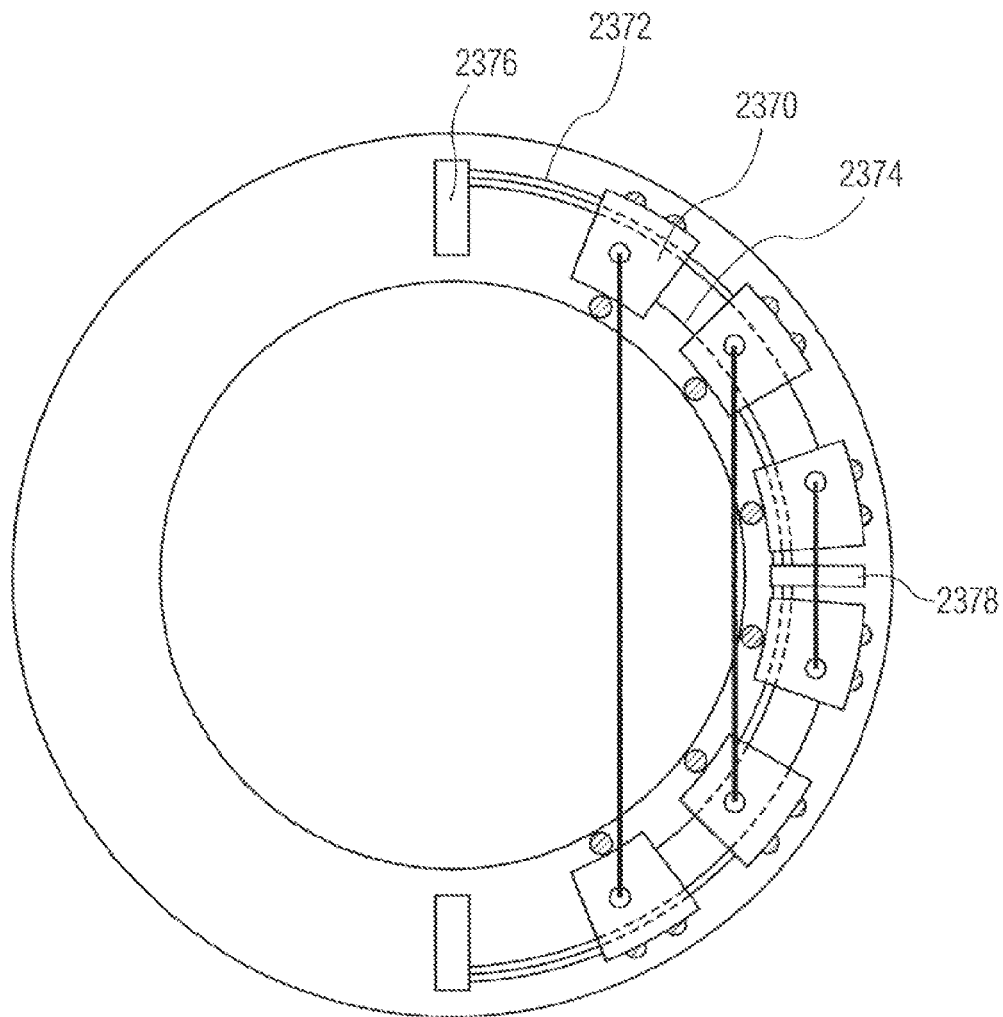


FIG. 23C

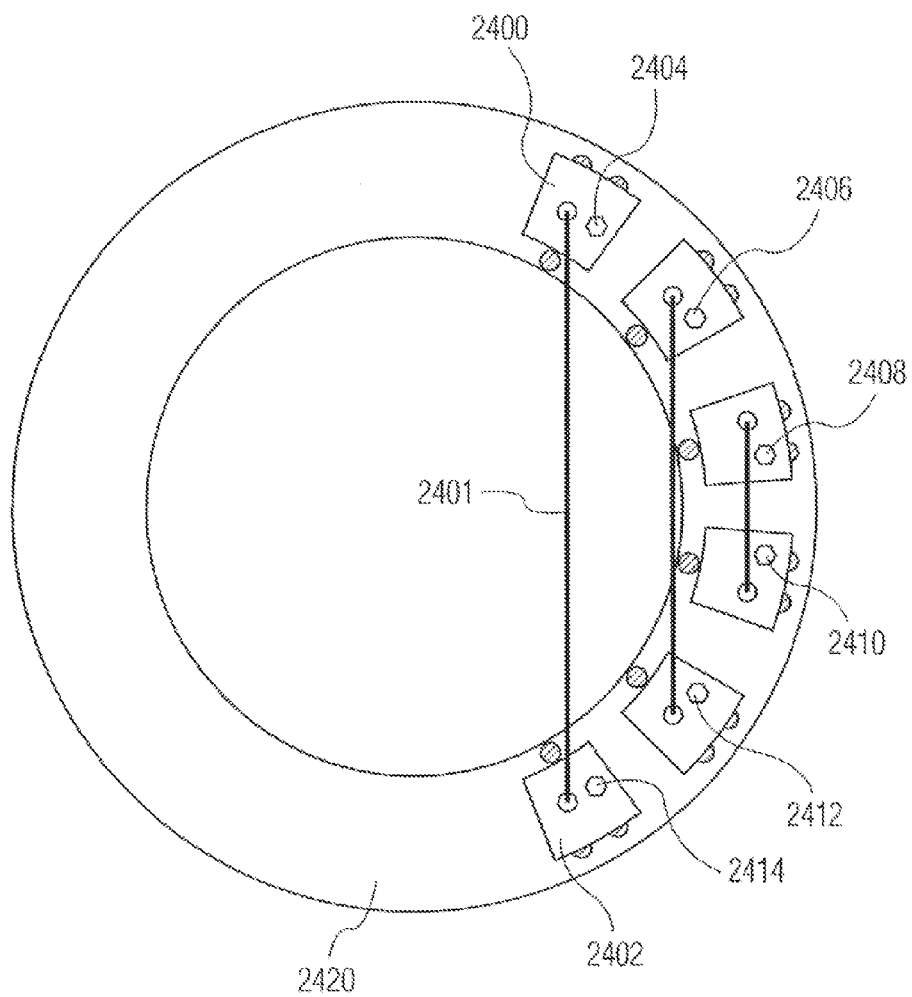


FIG. 24A

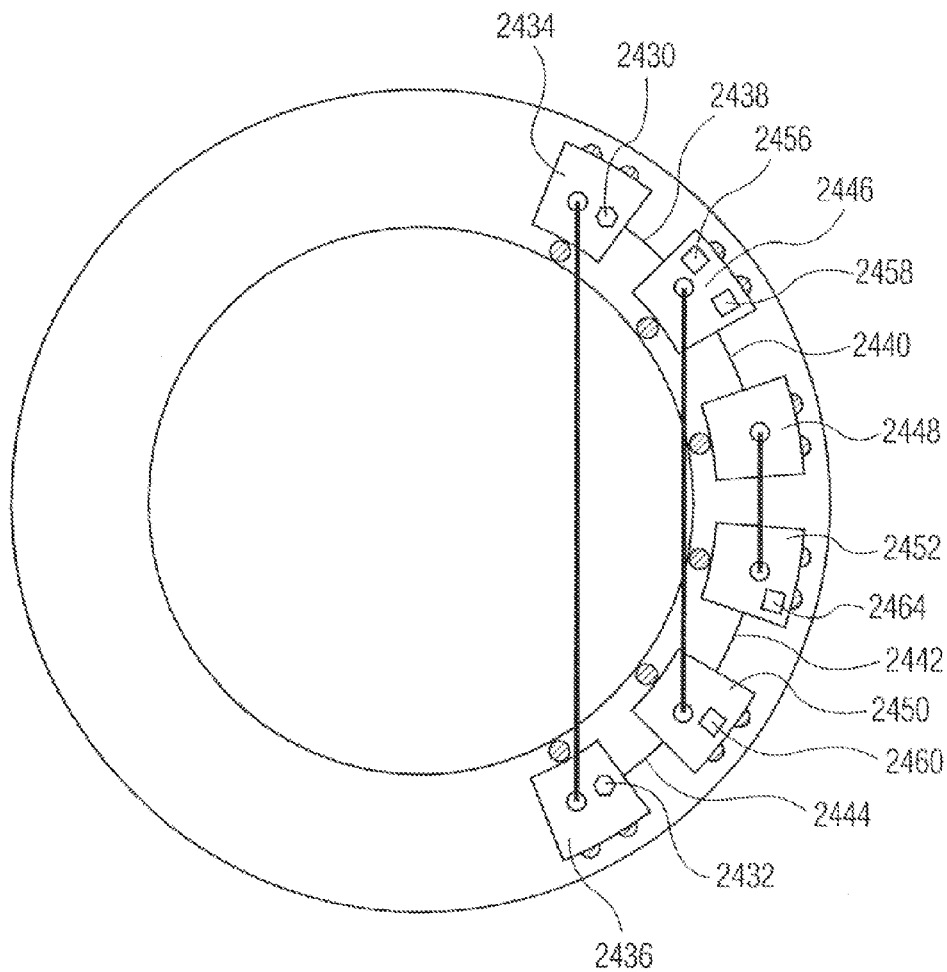


FIG. 24B

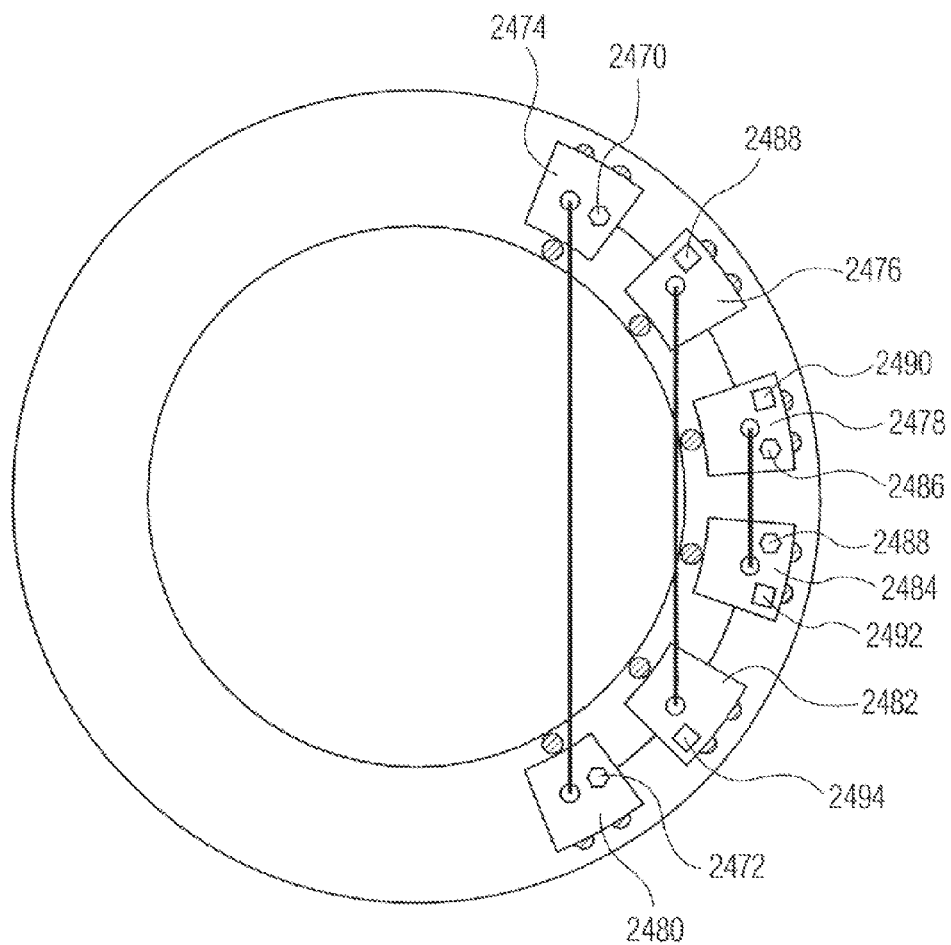


FIG. 24C

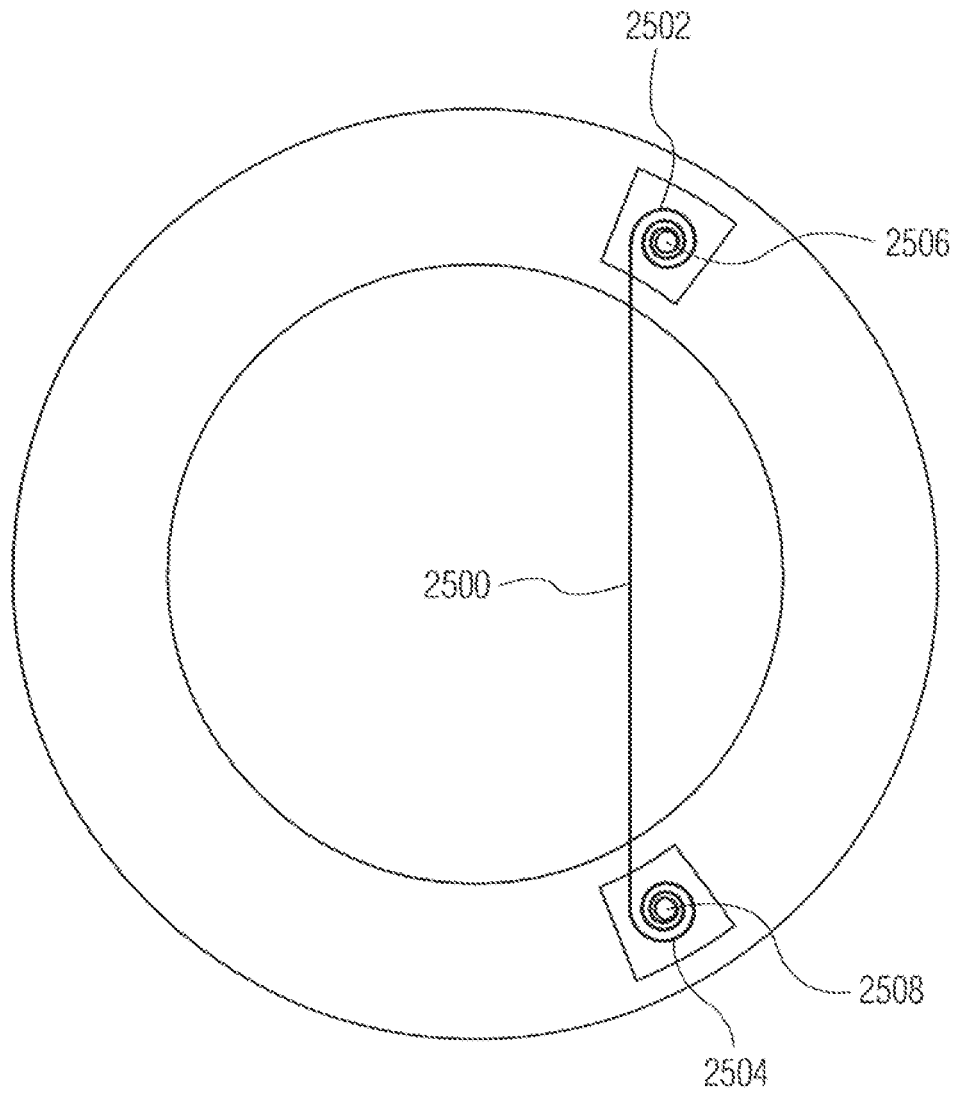


FIG. 25A

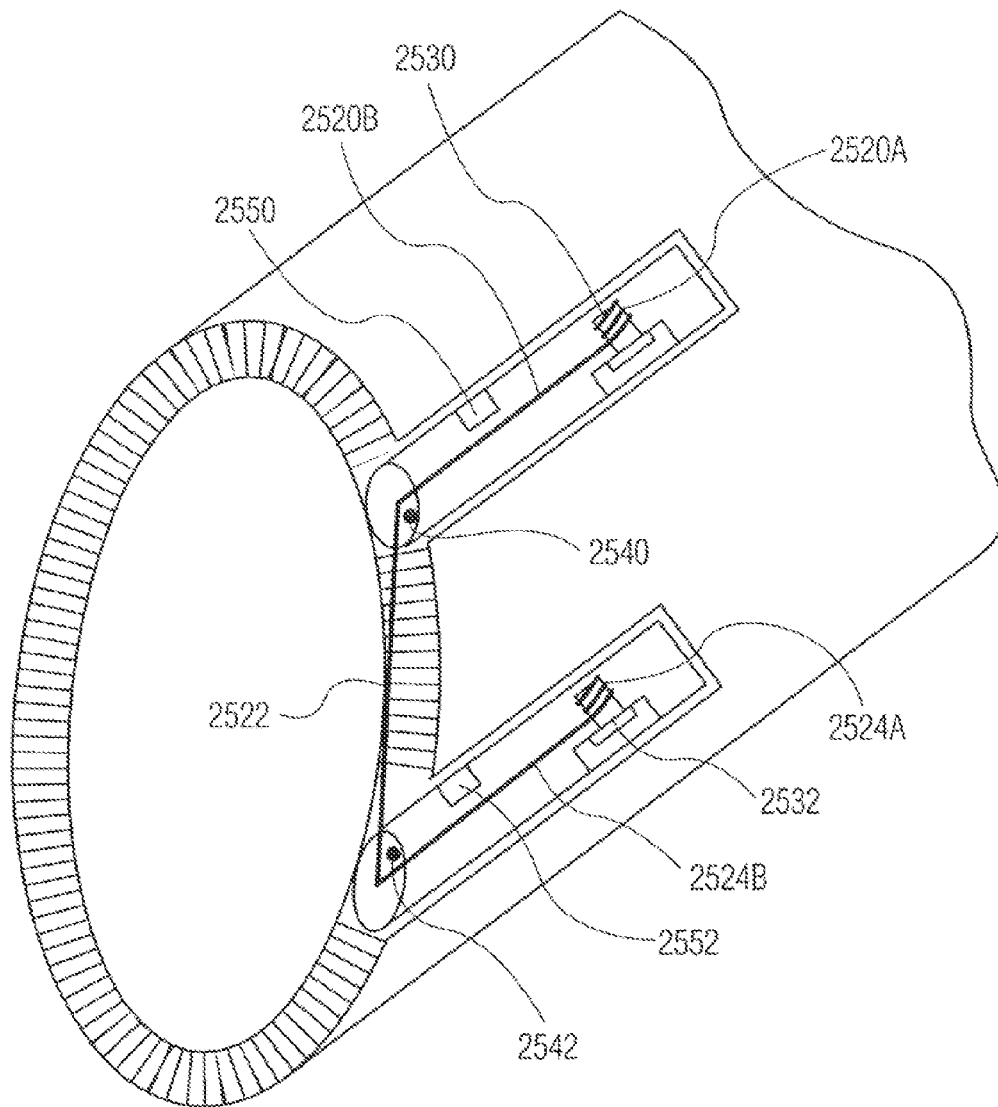


FIG. 25B

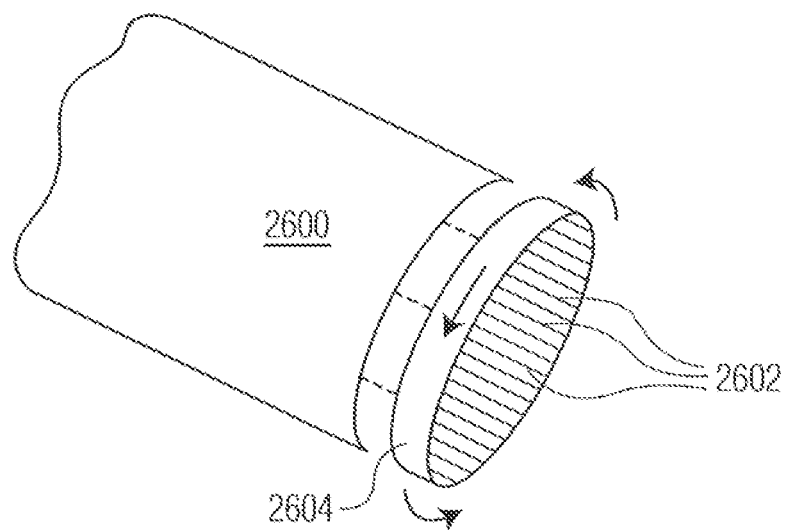


FIG. 26

1

RETRACTABLE BIRD AND DEBRIS DEFLECTOR FOR AN AIRCRAFT JET ENGINE

CROSS REFERENCE TO RELATED APPLICATIONS

The subject matter of this application is related to, and claims priority from, the following provisional and utility applications:

1) Provisional Application No. 61/205,381 filed Jan. 16, 2009, and 2) Provisional Application No. 61/205,785 filed Jan. 22, 2009; and

2) U.S. patent application Ser. No. 12/689,554, filed Jan. 19, 2010 (now allowed), from which this application is a continuation-in-part.]

Large sized debris which enters the intake of a jet engine may have disastrous consequences, including engine damage, functional engine destruction, and, if all or most engines become non-functional, emergency termination of a flight. This is what occurred on Jan. 15, 2009 with a flight out of LaGuardia Airport which made an emergency landing in the Hudson River after both of its engines failed: The source of damage was a flock of birds some of which entered the air intake of the engines, and rendered both engines non-functional.

U.S. Pat. No. 4,354,346 to Wooding discloses an intake duct for a jet engine which is not retractable. The engine intake extension of the invention is long and expected to be aerodynamically very demanding.

U.S. Design Pat. No. 433,029 to Eidson comprises a non-retractable cowl. Because it is non-retractable, it will exert aerodynamic inefficiencies throughout a flight.

U.S. Pat. No. 5,385,612 to Li discloses a cleaning system which is intended to be useful for jet engine intake. However, the device is not retractable, and is not able to provide jet air intake without very substantial aerodynamic limitation.

U.S. Pat. Nos. 4,137,535; 5,102,375 and 5,139,464 all relate to mechanisms for extending a telescoping antenna.

The subject matter of these prior U.S. patents is incorporated herein by reference.

The invention herein discusses methods and apparatus for preventing birds and other debris from doing damage to a jet engine using two types of deployable/protractible apparatus with acceptable aerodynamic features.

SUMMARY OF THE INVENTION

It is a principal object of the present invention to provide protection to an operating jet engine against airborne birds and other debris which may damage the engine.

It is a further object of the present invention to provide such protection using retractable apparatus, so that the aerodynamic consequences of such an apparatus are minimized, with respect to duration of use.

The invention herein discusses methods and apparatus for preventing birds and other debris from damaging a jet engine. It entails the deployment of a radially distributed set of first elements in front of the engine air intake. During the process of deployment, the leading edges of these first elements converge as they are extended from the engine housing. In order to prevent these first elements from suffering damage or malpositioning due to air turbulence, a second element, oriented transverse to the first elements, and positioned at the leading edge of the first elements, is also deployed. The second element features an adjustable circumference, allowing it to maintain the leading edges during the process of deployment,

2

with the circumference changing as the length of the deployed portion of the first element changes.

The first elements are retractable into the housing of the engine, so that once the aircraft rises above the altitude where such a strike may occur, better aerodynamic performance may be attained. During the landing phase of the flight, the first elements may be re-deployed when the aircraft has descended to an altitude where such protection is needed.

There are a variety of possible first element configurations involving variations in (a) the shape of the first element (straight and curved), (b) the number of first elements, and (c) the structural details of the first elements (for example: rigid rod terminating in eyelet, rigid rod terminating in tubular structure, hollow rod terminating in T-shaped tubular structure, and cable terminating in eyelet).

There are a variety of possible second element configurations involving variations in (a) the quality of the second element material (elastic, spring, cable), and (b) the number of second elements.

In one preferred embodiment of the invention, electromagnetic coupling secures adjacent leading edges of first elements in the fully deployed state.

in another preferred embodiment, de-icing apparatus warms the first and/or second elements.

In yet another preferred embodiment, the entire deflector apparatus rotates about the longitudinal axis, to provide additional protection.

Another embodiment of the invention entails the deployment of elongate linear elements which are deployed across the air intake duct of the engine, oriented perpendicular to the longitudinal axis of the engine. In the retracted state, these filter elements are moved to one or more sides of the intake duct, out of the incoming air stream. Guiding elements facilitate the deployment and retraction processes.

in a preferred embodiment of the invention, the elongate linear elements are a cable.

In another preferred embodiment of the invention more than one set of filter elements is deployed, with each set of elements having a different orientation.

In yet another preferred embodiment of the invention one or more sets of deflector elements rotates about a central longitudinal axis.

In yet another preferred embodiment of the invention a cleaning apparatus cleans the filter elements.

BRIEF DESCRIPTION OF THE DRAWINGS

FIG. 1A is a perspective view of a jet engine with a deployed deflector apparatus having multiple curved first elements and a single transverse second element.

FIG. 1B is a perspective view of a jet engine with a deployed deflector apparatus having multiple straight first elements and a single transverse second element.

FIG. 1C is a representational diagram of a jet engine indicating the storage of retracted first elements and a retracted second element.

FIG. 1D is a representational diagram of a jet engine indicating the storage of telescoping first elements.

FIG. 1E is a front view of the placement of first elements within an engine housing.

FIG. 2A is a representational diagram showing a front view of a fully deployed deflector apparatus with 16 first elements arrayed in a radially symmetric configuration, and a second element.

3

FIG. 2B is a representational diagram showing a front view of a partially deployed deflector apparatus with 16 first elements arrayed in a radially symmetric configuration, and a second element.

FIG. 2C is a representational diagram showing another front view of a partially deployed deflector apparatus with 16 first elements arrayed in a radially symmetric configuration, and a second element, showing a lesser degree of deployment than that shown in FIG. 2B.

FIG. 2D is a representational diagram showing another front view of a partially deployed deflector apparatus with 16 first elements arrayed in a radially symmetric configuration, and a second element, showing a lesser degree of deployment than that shown in FIG. 2C.

FIG. 2E is a representational diagram showing a front view of the deflector apparatus of FIG. 2D in a fully retracted state.

FIG. 3A shows a representational cross sectional diagram of a deflector apparatus with curved, telescoping first elements, and a second element, in a deployed position.

FIG. 3B shows a representational cross sectional diagram of a deflector apparatus with curved, telescoping first elements, and a second element, in a retracted position.

FIG. 4A shows a representational cross sectional diagram of a deflector apparatus with straight, telescoping first elements, and a second element, in a deployed position.

FIG. 4B shows a representational cross sectional diagram of a deflector apparatus with straight, telescoping first elements, and a second element, in a retracted position.

FIG. 5A shows a representation diagram of a deflector apparatus with four first elements, a second cable element, and a single winch for adjusting the length of the cable.

FIG. 5B shows a representation diagram of a deflector apparatus with four first elements, two second cable elements, and two winches for adjusting the length of the cables.

FIG. 5C shows a representation diagram of a deflector apparatus with four first elements, four second cable elements, and four winches for adjusting the length of the cables.

FIG. 5D shows a representational diagram of a T-shaped leading end of a first element, showing apparatus to decrease the friction due to motion of a cable.

FIG. 6A is a representational diagram showing a coiled second element, in a configuration corresponding to a fully deployed state.

FIG. 6B is a representational diagram showing a coiled second element, in a configuration corresponding to a partially deployed state.

FIG. 6C is a representational diagram showing a coiled second element, in a configuration corresponding to a partially deployed state, showing a lesser degree of deployment than that of FIG. 6B.

FIG. 6D is a representational diagram showing a coiled second element, in a configuration corresponding to a partially deployed state, showing a lesser degree of deployment than that of FIG. 6C.

FIG. 6E is a representational diagram showing a coiled second element, in a configuration corresponding to a fully retracted state.

FIG. 7A is a representational diagram showing a coiled second element passing through the leading edge of each of two T-shaped first elements, in a deployed configuration.

FIG. 7B is a representational diagram showing a coiled second element passing through the leading edge of each of two T-shaped first elements, in a retracted configuration.

FIG. 5A is a perspective view of a jet engine with a deployed deflector apparatus having multiple curved first elements and two transverse second elements.

4

FIG. 5B is a perspective view of a jet engine with a deployed deflector apparatus having multiple straight first elements and two transverse second elements.

FIG. 5C is a representational diagram of a jet engine indicating the storage of retracted first elements and two retracted second elements.

FIG. 9A is a representational diagram showing a front view of a fully deployed deflector apparatus with 16 first elements arrayed in a radially symmetric configuration, and two second elements.

FIG. 9B is a representational diagram showing a front view of a partially deployed deflector apparatus with 16 first elements arrayed in a radially symmetric configuration, and two second elements.

FIG. 9C is a representational diagram showing another front view of a partially deployed deflector apparatus with 16 first elements arrayed in a radially symmetric configuration, and two second elements, showing a lesser degree of deployment than that shown in FIG. 9B.

FIG. 9D is a representational diagram showing another front view of a partially deployed deflector apparatus with 16 first elements arrayed in a radially symmetric configuration, and two second elements, showing a lesser degree of deployment than that shown in FIG. 9C.

FIG. 9E is a representational diagram showing a front view of the deflector apparatus of FIG. 9D in a fully retracted state.

FIG. 10A is a representational diagram showing a cross sectional view of a tubular T-shaped first element, with projections forming two pairs of second elements, containing cables.

FIG. 10B shows a representation diagram of a deflector apparatus with four first elements, four second cable elements each located at the leading edge of the first elements, four additional second cable elements each located between the leading edge and the trailing edge of the first elements, and four winches for adjusting the length of the additional cables elements.

FIG. 10C is a representational diagram of a jet engine indicating the storage of retracted first elements and two retracted second elements.

FIG. 11A is a perspective view of a jet engine with a deployed deflector apparatus having multiple curved first elements and six transverse second elements.

FIG. 11B is a perspective view of a jet engine with a deployed deflector apparatus having multiple straight first elements and six transverse second elements.

FIG. 12A is a representational diagram showing a front view of a fully deployed deflector apparatus with 16 first elements arrayed in a radially symmetric configuration, and six second elements.

FIG. 12B is a representational diagram showing a front view of a partially deployed deflector apparatus with 16 first elements arrayed in a radially symmetric configuration, and six second elements.

FIG. 12C is a representational diagram showing another front view of a partially deployed deflector apparatus with 16 first elements arrayed in a radially symmetric configuration, and six second elements, showing a lesser degree of deployment than that shown in FIG. 12B.

FIG. 12D is a representational diagram showing another front view of a partially deployed deflector apparatus with 16 first elements arrayed in a radially symmetric configuration, and six second elements, showing a lesser degree of deployment than that shown in FIG. 12C.

FIG. 12E is a representational diagram showing a front view of the deflector apparatus of FIG. 12D in a fully retracted state.

5

FIG. 13 is a representational diagram of a deflector apparatus with 40 T-shaped first elements in a fully deployed configuration.

FIG. 14 is a representational diagram of two T-shaped first elements with electromagnetic apparatus at two adjacent projections.

FIG. 15A is a representation diagram of a deflector apparatus with four cable-based first elements, four winches for adjusting the length of the respective cables, and a cable-based second element associated with a tubular T-shaped additional first element and with an additional associated winch.

FIG. 15B is a perspective view of a jet engine with a deployed deflector having the apparatus shown in FIG. 15A.

FIG. 16A is a cross sectional view of a portion of a hinge and a hinge-controlling apparatus for attaching a first element to a jet engine, showing a deployed state of the first element.

FIG. 16B as a cross sectional view of the hinge and hinge-controlling apparatus of FIG. 16A, showing a transitional state between the deployed state and the retracted state.

FIG. 16C is a cross sectional view of the hinge and hinge-controlling apparatus of FIG. 16B, showing the retracted state.

FIG. 17 is a perspective view of a jet engine with a deployed deflector apparatus having multiple straight first elements and a single transverse second element, with the deflector apparatus showing rotational motion about the longitudinal axis of the engine.

FIG. 18A shows a representational oblique view of a jet engine.

FIG. 18B shows a representative oblique view of a jet engine with a fully deployed bird and debris filter comprising two sets of elements.

FIGS. 19A-19E shows a schematic of the gradual deployment of a bird and debris filter with two groups of elements.

FIGS. 20A-20E shows a schematic of the gradual deployment of a bird and debris filter with one group of elements.

FIGS. 20F-20J shows a schematic view of a mechanism for providing filter elements with multiple orientations without the use of a second filter.

FIG. 21 shows another oblique view of a jet engine with a deployed set of two filters, with elements of one filter aligned in a different direction than that of the elements of the other filter.

FIGS. 22A and 22B show a schematic frontal view of both filter elements and guiding elements, in the retracted and deployed states, respectively.

FIGS. 23A-23C show schematic views of exemplary means for causing the positioning of the guiding elements of FIGS. 22A and 22B.

FIGS. 24A-24C show schematic views of exemplary means for causing the positioning of the guiding elements of FIGS. 22A and 22B.

FIGS. 25A and 25B show schematic views of means for releasing a portion of the filter elements during the process of deployment, and taking up a portion of the filter elements during the process of retraction; and show a cleaning apparatus.

FIG. 26 shows a schematic oblique diagram in which a group of filter elements is operative to rotate about an axis parallel to that of the long axis of the engine.

DESCRIPTION OF THE PREFERRED EMBODIMENT

FIGS. 1A and 1B show two types of deflector apparatus for a jet engine. FIG. 1A shows 12 curved first elements 102

6

projecting from the air intake end of jet engine 100. FIG. 1B shows 11 straight first elements 104 projecting from the air intake end of jet engine 101. Many other configurations are possible. Both figures show first elements in a radially symmetric distribution. Configurations with a greater or lesser number of first elements are possible.

To stabilize the first elements during deployment, a second expandable element connects the leading end of the first elements, shown as 103 in FIGS. 1A and 105 in FIG. 1B.

Embodiments of the invention in which the first elements link at a point or a small area without an aperture (i.e. an opening at the most forward point) are possible. Embodiments of the invention in which the circular apparatus is substituted by an apparatus of another shape are possible. Elliptical shapes, hexagonal shapes, octagonal shapes, polygonal shapes (and any shape in which the number of sides to the central aperture is equal to the number of first elements) are possible. A shape which is identical to that of the engine housing, if not circular, is possible.

FIG. 1C shows the first elements 108 in the fully retracted position (indicated in the figure as broken lines) within the housing of engine 106. In the embodiment shown in FIG. 1C, the first elements in the retracted state are stored inside the engine housing, and are not telescoped. The configuration shown in FIG. 1C would be suitable for straight first elements, and could also be used for curved first elements with a large radius of curvature. Curved first elements with a smaller radius of curvature (i.e. approximately the same radius of curvature as the engine housing) could be stored by rotating them 90 degrees so that they would, in the stored state, lie along the circumference of the engine.

FIG. 1D shows the storage of telescoped first elements 110. The advantage of telescoping is ease of storage in the retracted state. The telescoped configuration for the retracted state shown in FIG. 1D could also accommodate curved first elements (e.g. as shown in FIGS. 3A and 3B) with a smaller radius of curvature (without having to rotate 90 degrees for storage) than that of the engine housing of FIG. 1C.

FIG. 1E shows a front view of telescoped first elements 114 within compartments 112, situated in engine housing 118. The engine apparatus is situated centrally, in the location indicated by 116.

FIGS. 2A through 2E show front views of an engine with the first elements joined at the leading edge by a circular second element. During the deflector retraction process, the circular second element increases in circumference as shown in the sequence of FIGS. 2A to 2E. FIG. 25 shows the fully retracted state—where most or all of the first element apparatus has been withdrawn into the engine housing, as the radius of what was the central circular element is increased to the point that it equals approximately that of the engine housing. A 16 first element configuration is shown; configurations with fewer and larger numbers of first elements are possible.

FIGS. 3 and 4 show a side view of an embodiment of the apparatus, emphasizing a first element structure with a telescoping configuration. FIG. 3 shows an apparatus with curved first elements, and FIG. 4 shows an apparatus with straight first elements. Each figure shows that the first element structure consists of a series of cylindrical elements with a telescoping structure, much like a retractable automobile antenna. FIG. 3A shows the first elements 300A fully extended, with circular stabilizing apparatus 310A assuming a minimal circumference. In the fully retracted state shown in FIG. 3B, the telescoping of concentric cylindrical first elements 300B allows the first elements to fit inside the engine housing, while the circular apparatus 310B expands (in a process described hereinbelow) so that it may also fit into the

engine housing. Only two sets of first elements are shown for simplicity. Configurations with more sets of first elements are desirable to allow for greater stability and ease of retraction. Arrows above FIGS. 3A and 3B show the direction of air flow into the engine. Electrically controllable telescoping elements which are controlled by one or more electric motors, by hydraulic apparatus and by pneumatic apparatus are known.

FIG. 4A shows the first elements 400A fully extended, with circular stabilizing apparatus 410A assuming a minimal circumference. In the fully retracted state shown in FIG. 4B, the telescoping of concentric cylindrical first elements 400B allows the first elements to fit inside the engine housing, while the circular apparatus 410B expands (in a process described hereinbelow) so that it may also fit into the engine housing. Only two sets of first elements are shown for simplicity. Configurations with more sets of first elements are desirable to allow for greater stability and ease of retraction. Arrows above FIGS. 4A and 4B show the direction of air flow into the engine.

FIGS. 5A-5C show a possible structure for controlling the configuration of the second element. Apparatus with only four first elements is shown for ease of interpretation. Referring to FIG. 5A, as first elements 510A-D are retracted (by movement which is radially outward in the figure), cable 512, the second element, is gradually unspooled from spool 518 by motorized apparatus 516 in housing 514. (The segment of cable which lies between each of 510A/B, 510B/C, 510C/D and 510D/A is shown in a curved configuration, which would be the conceptual limiting case with an infinite number of first elements; In the case of a large number of first elements, a many-sided polygon would approximate the circle shown in the figure.) As shown in the figure, the terminal segment of each radial arm forms a curved cylindrical shape which allows the cable to pass through. When the first elements go from the retracted state to the deployed state, motor 516 causes cylinder 518 to take up cable slack as it becomes available. Embodiments of the invention in which 512 is a spring apparatus (see hereinbelow) are possible.

FIG. 5S shows an apparatus similar to that of 5A, except that there are two cables, 542A and 542B. The slack for one end of each of 542A and 542B is controlled by slack uptake and release apparatus 544A and 544B (each of which operate in similar fashion to that of 514).

FIG. 5C shows an apparatus similar to that of FIGS. 5A and 5B except that there is one cable segment (572A-D) for each pair of adjacent retractable arms, and one slack uptake and release apparatus (574A-D) for one end of each pair of adjacent cable ends. For example, when the apparatus in FIG. 5C goes from the deployed to the retracted state:

574A unrolls appropriate portions of one end of cable 572A and one end of cable 572B; 574B unrolls appropriate portions of one end of cable 572B and one end of cable 572C;

574C unrolls appropriate portions of one end of cable 572C and one end of cable 572D; and 574D unrolls appropriate portions of one end of cable 572D and one end of cable 572A.

Configurations of the invention with various friction reducing elements are possible. FIG. 5D shows a representational cross sectional view of the terminal protuberance of a first element in one embodiment of the invention. In order to minimize friction between the cable and the first element, one or more of friction reducing elements 592A, 592B, 594A, 594B, 596A, 596B, 598A, and 598B are included. These may be flat rolling elements, or grooved wheels. In another embodiment of the invention, 592A and B may be considered to be a cross-sectional representation of a circular bearing

device which guides the cable with minimal friction; the same is true of the 594A and B pair, the 596A and B pair and the 598A and B pair. Embodiments of the invention with either a greater or lesser number of guiding elements are possible. Embodiments of the invention in which the friction-reducing elements are actively lubricated, or are self-lubricating are possible. Many other friction reducing configurations will be apparent to those skilled in the art.

FIGS. 6A to 6E, shows a circular spring apparatus which provides the attractive force between adjacent distal ends of the first elements when they are deployed. FIGS. 6A to 6E correspond, respectively to the states of deployment/retraction shown in FIGS. 2A to 2E, i.e., five states ranging from first elements fully deployed (FIG. 6A) to first elements fully retracted (FIG. 6E).

FIG. 7, consisting of FIGS. 7A and 7B shows the circular spring apparatus of FIG. 6 in conjunction with first elements in two different states of retraction/deployment:

FIG. 7A corresponds to FIGS. 2B and 6B; while

FIG. 7B corresponds to FIGS. 2D and 6D.

Embodiments of the spring apparatus shown in FIGS. 6 and 7 in which one end of the spring is anchored to a first element are possible.

FIGS. 8A and 8B (analogous to FIGS. 1A and 1B respectively) show a configuration of the apparatus in which there are two transverse/second element supporting apparatus structures (810 and 820 for engine 815 in FIG. 8A, and 830 and 840 for engine 835 in FIG. 8B). The mode of operation of the transverse apparatus shown in each of the two figures is similar to that of the configurations with one transverse supporting apparatus, i.e. in the case of two such apparatus, each is retractable as shown by 850 and 860 in FIG. 5C. In FIG. 5C, the engine is indicated by 855 and the first support elements are indicated by broken lines 870.

FIGS. 9A-9E (each showing two circular second elements) are analogous to FIGS. 2A-2E (each showing one circular second element). As indicated hereinabove, in many configurations the circle representing the second element in the figure represents the theoretical upper limit of a many-sided polygon.

In FIG. 9A, the fully deployed configuration, the distal (i.e., nearest to the leading end) circular apparatus is 910A and the proximal (i.e. nearest to the trailing edge) one is 920A. In FIG. 9B, the partially retracted configuration, the distal circular apparatus is 930B and the proximal one is 940B. In FIGS. 9C-E, the proximal circular apparatus is retracted within the engine housing, so the appearance is identical to FIGS. 2C-2E, respectively.

FIG. 10A shows a representational view of a complex first element for a configuration with one distal transverse supporting apparatus and one transverse supporting apparatus in the mid-portion of the first element. In principle, such a complex element has features of both a first element (i.e. as it exits the engine housing, one section extends longitudinally) and a second element (has projections which, as they exit the housing, extend in a direction transverse to the longitudinal section). The entire apparatus of FIG. 10A is analogous to element 510A in FIG. 5A (which is a first element without a transverse supporting apparatus in its midportion). Referring again to FIG. 10A, cables 1000A and 1000B help to align and hold together the distal ends of the complex first elements; They are analogous to any of [a] 512 in FIG. 5A; [b] 542A and 542 in FIG. 5B; and [c] any of (i) 572A and 572B, (ii) 572B and 572C, (iii) 572C and 572D, and (iv) 572D and 572A in FIG. 5C. In addition, cables 10000 and 1000D serve to anchor the midportion of each complex first element.

When the apparatus in FIG. 10A is utilized in a configuration analogous to that of FIG. 5C (i.e. an array of four of complex first element 1002), each cable segment analogous to 1000C in FIG. 10A extends to the neighboring first element to the left (see FIG. 10B), enters its main shaft, and comprises the segment analogous to 1000D in that left neighboring first element. Similarly, each cable segment analogous to 1000D in FIG. 10A extends to the neighboring first element to the right (see FIG. 10B), enters its main shaft, and comprises the segment analogous to 1000C in that right neighboring first element.

In a configuration analogous to that of FIG. 5A, the cable segment 1000C would extend from shaft 1002, out through projection 1004, and thence through the midportion of each first element (via projections analogous to each of 1004 and 1006), and ultimately return to first complex element 1002 via projection 1006 to form cable segment 1000D.

In configurations analogous to that shown in FIG. 5B, the cable segment 1000C would extend from shaft 1002, out through projection 1004, through the midportions of two or more adjacent first elements, each first element (via projections analogous to each of 1004 and 1006), and ultimately enter another first element via a projection analogous to 1006, and form a cable segment analogous to segment 1000D in another first element.

In configurations analogous to that of FIG. 5B, the cables within first elements traversed by transverse cable segment must be geometrically set up so that the transverse segment does not contact longitudinal segments analogous to 1000A and 1000B. Although FIG. 10A shows all four cable segments lying in the same plane (i.e. the plane of the figure), in three dimensions, the transverse segment could cross through the shaft either so that it does not contact either of the segments analogous to 1000A or 1000B (i.e. by crossing above or below the plane defined by 1000A and 1000B).

The advantage of projections 1004 and 1006 is that they help guide and secure the transverse cable in the midportion of the first element, and allow for a locking mechanism to further stabilize the apparatus. The disadvantage is that they add weight, they further restrict the open area in front of the engine, and they make retraction of the first element more complex. Embodiments of the invention in which each of projections 1004 and 1006 are absent, replaced by respective openings in the shaft of 1002 to accommodate respective cables 1000C and 1000D are possible.

Though FIG. 10A shows projections 1004 and 1006 to be in the midportion of the first element, configurations are possible in which the junction is asymmetrically located, either proximally (i.e. nearer to the engine housing) or nearer to the distal end of the apparatus.

The proximal ends of each of 1000C and 1000D are linked to cable control apparatus which appropriately releases or takes in cable, as the situation may require. Such cable control apparatus is analogous to any of [a] 514, 516 and 518 shown in FIG. 5A; [b] 544A and 544B shown in FIG. 5B; and [c] 574A, 574B, 574C and 574D shown in FIG. 5C.

Cables may be secured within 1002 by a variety of means and mechanisms including:

a) situating the cable within a non-moving sheath; b) grooves within 1002 for each cable; and/or c) one or more guiding wheels, rollers, or bearings along the length of the cable within 1001, 1002, 1003, 1004 and/or 1006, analogous to that which is shown herein in conjunction with FIG. 5D.

FIG. 10B shows a deflector which includes an array of four of the complex first elements shown in FIG. 10A. The apparatus shown in the figure is analogous to that shown in FIG. 5C. However, the apparatus in FIG. 10B includes an addi-

tional transverse support group of structures. Cable take-up apparatus 1008 controls the length of cable segment 1000D, which passes through 1002, exits through projection 1006 and enters the corresponding structure on the right side of the figure. Similarly, cable take-up apparatus 1008 controls the length of cable segment 1000C, which passes through 1002, exits through projection 1004 and enters the corresponding structure on the left side of the figure. The operation of 1008 and associated components is similar to that of 574A-D in FIG. 5C. These aforementioned structures link the midsection (which need not be located at the geometric middle) of the complex first elements (e.g. 1002).

The cable segments which forms the distal second element exit through projection 1003 as 1000B, and then enters the corresponding structure indicated by elements on the right side of the figure; Another cable segment which forms the distal second element exit through projection 1001 as 1000A, and then enters the corresponding structure indicated by elements on the left side of the figure.

The four cable take-up apparatus for the distal second elements is not shown in the figure, but is similar to that of 1008, and 574A-D. Long broken lines in the figure indicate cable for the distal/leading edge second elements which are contained within 1002. Although these cable segments extend into the proximal shaft of 1002 (as shown in FIG. 10A), these segments of cable are not shown in the figure, for clarity. Short broken lines indicate cable for the proximal/midportion second elements, which are shown in their full extent.

FIG. 10C, analogous to FIG. 1D, shows a representational view of the retracted state, of an embodiment with (a) one transverse stabilizing cable 1020 in its midportion, and (b) collapsible/telescoping first elements 1022. With embodiments of the invention with lateral protuberances in the midsection, there will be a limitation to the collapse above and below such midsection protuberances. An embodiment of the invention is also possible in which the midsection protuberances themselves are able to collapse/telescope.

The telescoped configuration for the retracted state shown in FIG. 10C could also accommodate curved first elements (e.g. as shown in FIGS. 3A and 3B), as discussed hereinabove in conjunction with FIG. 1D.

Whereas the aforementioned embodiments contain either no transverse elements along the first elements, or one such element (FIGS. 5A to 9E), FIG. 11A shows a configuration with 5 transverse elements and curved first elements (analogous to FIGS. 1A and 8A) and FIG. 11B shows a configuration with 5 transverse elements and straight first elements (analogous to FIGS. 1B and 8B). Configurations with greater and lesser numbers of first elements are possible. More first elements result in a greater degree of first element stability and the ability to limit the maximum size of an object which may cross the barrier resulting from the deployment of the apparatus described herein. On the other hand, more first elements result in greater weight, greater resistance to air entry and more complex cable arrangements within first elements and more complex cable supporting apparatus.

FIGS. 12A-12E—analogous to FIGS. 2A-2E and 9A-9E—show a front view of some of the successive steps in the transition from a fully deployed apparatus (FIG. 12A) to a fully retracted one (FIG. 12E) for a configuration with five transverse elements, 1200, 1202, 1204, 1206 and 1208 (in addition to the distal transverse support common to all of the configurations hereinabove). FIG. 12B shows a state in which two of the five transverse elements have been retracted (and in which the non-retracted transverse elements and the distal supporting apparatus have each (i) been pulled back and (ii) undergone an increase in radius). FIG. 12C shows a state in

11

which four of the five transverse elements have been retracted (and in which the one remaining non-retracted transverse element and the distal supporting apparatus have each (i) been further pulled back and (ii) undergone a further increase in radius). FIG. 12D shows a state in which all of the five transverse elements have been retracted (and in which the remaining non-retracted distal supporting apparatus has (i) been still further pulled back and (ii) undergone a still further increase in radius).

FIG. 13 shows a front view of a fully deployed engine protection device with 40 first elements (1300A), in which first element has a terminal protuberance (1300B) which is analogous to 1001 and 1003 of FIG. 10A herein. Cable or cables 1302, analogous to the cable shown in any of the configurations of FIGS. 5A, 5B and 5C, serve to draw the protuberances together as the device is deployed, and to stabilize the protuberances as the device is retracted. In addition 1302 may secure each of the protuberances 1300B so that they are in secure contact with each other. Another mechanism for securing each 1300 to its two adjacent neighboring 1300Bs is to have the surface of each form a secure fit with its neighboring 1300B, either because the surfaces are parallel, or because the surfaces have complementary extensions and depressions which promote such a fit. Furthermore, by making the projections and depressions cone-shaped rather than cylindrical, a non-perfect alignment of adjacent first elements during deployment may be corrected for.

In another embodiment of the invention, a magnetic attraction between adjacent protuberances may be used to promote their attraction during deployment. The magnetic mechanism may be from fixed elements (e.g. one side of each protuberance is a north magnetic pole, and the other side is a south pole, such that the arrangement is:

... (N-S)-(N-S)-(N-S)-(N-S) ...

Alternatively, the source of magnetism may be electromagnetic, as shown in FIG. 14, thereby allowing for a simple means of turning off the attractive mechanism. FIG. 14 shows a coil of conducting wire 1404A on one end of first element 1400A for generating a magnetic field when a current is passed through it. The wires need not be on the surface of the object, and may be embedded beneath the surface. The ends of the coil 1404B pass through the shaft of 1400A to a power supply and control unit. There is corresponding apparatus 1406A on the end of first element 1402A for generating a magnetic field when a current is passed through it. The ends of the coil 1406B pass through the shaft of 1402A to a power supply and control unit. The orientation and winding of the coils is such that 1404A attracts 1406A when a current is passed through each. In a preferred embodiment, additional coils are placed symmetrically on each projection, i.e. 1400B and 1402B, to allow for the attraction to each of their respective neighboring projections.

In yet another embodiment of the invention, an active locking mechanism between adjacent protuberances is possible. Activation and deactivation of the locking mechanism may be electric or via one or more cables which traverse one or more of first elements with such a mechanism.

FIG. 15A shows an embodiment of the invention in which the first elements are not composed of rigid rods. These first elements consist of cables 1502A-D. At their respective proximal ends are cable take-up and release apparatus 1500A-D; At their respective distal ends is an eyelet 1504A-D, which allows each of 1502A-D to be pulled during the deployment process. Deployment is caused when cable take-up 1506 winds in 1510, causing the perimeter of this cable loop to decrease. As the decrease occurs cables 1502A-D are pulled out of 1500A-D. The tension on the loop 1510 exerted

12

by each of 1500A-D is adjusted to keep loop 1510 centered over the air intake. In one version of this embodiment of the invention, an apparatus 1520 (either electromechanical, hydraulic or pneumatic) pushes 1508 distally (toward the center of the air intake) during deployment. The retraction of the deflector involves active uptake of cables 1502A-D by take-up apparatus 1500A-D, with simultaneous spooling out of cable from 1506. In the version which includes 1520, it may be used to facilitate the retraction of 1508. The tension of each of 1500A-D on each respective one of 1502A-D is adjusted, during the retraction process, to keep the deflector properly centered at all times.

FIG. 15B shows a perspective view of a jet engine 1530, and the first elements and second elements (with element numbers corresponding to those of FIG. 15A) which make up this embodiment. The embodiment shown in the figure contains no rigid support elements except for 1508. It would therefore be situated at the mouth of the engine.

Versions of this embodiment with two or more sets of apparatus to shorten loop 1510 are possible. Versions are also possible in which each of 1502A-D is a rigid telescoping rod, anchored to the engine housing, and deployed by the force exerted by cable take-up device 1506.

FIGS. 16A-16C show an embodiment of a hinge which anchors a first element 1600 to the engine housing, and is retractable. The first element is joined to one hinge component 1602, and retraction rod 1608 is joined to the other hinge component 1606. 1602 and 1606 pivot about 1604. 1608 is moved in and out by apparatus 1610, either mechanically or electromagnetically. 1608 is anchored to inner housing wall 1612 (anchoring not shown in figure), which is contiguous with 1614 which is the support apparatus for the engine.

FIG. 17 shows an embodiment of the invention in which the first and second elements apparatus rotate along the long axis of the engine, thereby to reduce the aerodynamic consequences of a fixed first element configuration, to reduce asymmetric engine wear, and to more efficiently deflect debris and/or birds. In the figure, the base of the deflector apparatus 1702 is contiguous with engine 1700, but is able to rotate about the long axis of the engine.

Embodiments of the inventions hereinabove are possible in which:

- 1) There is more than one distal cable running around the circumference of the device, to impart additional stability;
- 2) There are two or more cables running in parallel through the transverse/non-distal second elements (one cable illustrated hereinabove);
- 3) The cable is replaced or supplemented by one or more ribbon shaped elements;
- 4) There are two tandem deflector apparatuses, each of which has the appearance of all of the protection elements shown in FIG. 11A (or 11B, 1A, 1B, 8A or 8B). In a preferred embodiment of the invention, the first elements of the first apparatus are placed so that debris which passes through the outer apparatus is geometrically unlikely to pass through the second apparatus. The longitudinal first elements of the outer apparatus may have a different angular location than those of the inner apparatus, and/or the transverse elements of the outer apparatus may be situated in a more (or less) distal location than those of the inner apparatus. The outer apparatus may rotate (a) at a different speed than the inner one; and/or (b) in a different direction than the inner one;
- 5) The arrangement of first elements functions to (a) deflect airborne debris, and/or (b) break up airborne degree into smaller pieces.
- 6) Embodiments of the invention with other first element retraction and extension mechanism are possible.
- 7) Embodiments of the invention with other stabilizing mechanisms for the distal end of the first elements are possible.
- 8) Embodiments of the

13

invention with a device, such as a device for passing electric current through the deflector elements, for maintaining the temperature of the elements above freezing, thereby to prevent formation of ice on the deflector.

The retractable bird and debris filter described heretofore consists of elements which project forwards from the jet engine. A second type of retractable bird and debris filter is comprised of elements largely confined to the vicinity of a plane defined by the forward opening of the engine, described hereinbelow.

FIGS. 18A and 18B show a jet engine with such a bird and debris filter in the retracted, and in the deployed state, respectively.

FIG. 18A shows a perspective view of a jet engine 1800, with air intake shown on the right side of the figure. The filter is in the retracted state, and is not seen in the figure.

FIG. 18B shows a perspective view of engine 1802, with deployed filter 1804A. The filter shown in the figure has two sets of parallel linear filter elements, with one set of elements 1804B oriented perpendicular to the other set 1804C. Embodiments of the invention with one set, and with three or more sets of filter elements are possible. Embodiments of the filter in which one filter element is neither perpendicular nor parallel to another filter element are possible. In the embodiment of the filter shown in the figure, the filter elements form a grid over the air intake of the engine.

Clearly, increasing either (a) the number of filter elements or (b) the thickness of the elements, the greater the impedance to engine air intake. On the other hand, small numbers of filter elements or excessively thin elements will decrease the effectiveness of the filtration process.

The filter elements may be metallic, may be composed of a non-metal, or may be a composite of metallic and non-metallic elements. Each filter element may comprise a single strand of material or multiple strands comprising a cable. The strands, if multiple, may or may not be twisted or braided. The filter elements may or may not have elastic properties. Other filter element configurations will be apparent to those skilled in the art.

FIGS. 19A-19E show a schematic view of an example of the process of deployment of a set of vertically oriented filter elements, as seen looking into the engine from the air-intake side. FIG. 19A shows the retracted state of the filter, i.e. no filter elements are seen. The engine perimeter is indicated by 1920.

In FIG. 19B, the beginning of the filter element deployment process, one filter element 1900 extends, vertically oriented, across the left hand portion of the air intake, having moved from a storage location (not shown in this figure) at the extreme left of the figure. Another filter element 1901 extends, vertically oriented, across the right hand portion of the air intake, having moved from a storage location (not shown in this figure) at the extreme right of the figure.

FIG. 19C shows further progression of the deployment process beyond that shown in FIG. 19B:

On the left side of the figure, filter element 1900 has moved further to the right, and additional filter element 1902 has begun to traverse the intake opening; and

On the right side of the figure, filter element 1901 has moved further to the left, and additional filter element 1903 has begun to traverse the intake opening.

FIG. 19D shows another step in the progression of the deployment process beyond that shown in FIG. 19C:

On the left side of the figure, filter elements 1900 and 1902 have moved still further to the right, and additional filter element 1904 has begun to traverse the intake opening; and

14

On the right side of the figure, filter elements 1901 and 1903 have moved still further to the left, and additional filter element 1905 has begun to traverse the intake opening.

FIG. 19E shows yet another step in the progression of the deployment process, beyond that shown in FIG. 19D:

On the left side of the figure, filter elements 1900, 1902 and 1904 have each moved still further to the right, and additional filter element 1906 has moved to extend across a portion of the intake opening; and

On the right side of the figure, filter elements 1901, 1903 and 1905 have moved still further to the left, and additional filter element 1907 has moved to extend across a portion of the intake opening.

This exemplary figure shows a total of eight filter elements. Embodiments of the invention with a greater or lesser number of elements is possible. In the figure, in the deployed state, spacing between the elements is seen to be roughly equal. Embodiments of the invention in which the spacing is not equal are possible. The four step deployment process shown by FIGS. 19A to 19E is not intended to indicate that deployment is a step-wise process; it may be stepwise, or continuous; and if continuous the elements may move from the retracted to the deployed position at a constant or non-constant speed.

The relative width of the filter elements in the figure is not intended to indicate an actual relative width. The width may vary from element to element among an array of such elements. The width may vary along the length of an individual filter element. The vertical orientation of the filter elements is exemplary, and embodiment of the invention with horizontally oriented elements are possible, as well as embodiments in which the elements are neither vertical nor horizontal. Furthermore, embodiments of the invention are possible in which the filter elements are not parallel to each other—either during the process of deployment or in the fully deployed state.

FIGS. 20A-20E show a schematic view of an example of the process of deployment of a set of vertically oriented filter elements, as seen looking into the engine from the air-intake side, in which all of the elements are stored in a single group (FIGS. 19A-19E having shown the case of two stored groups). FIG. 20A shows the retracted state of the filter, i.e. no filter elements are seen. The engine perimeter is indicated by 2020.

In FIG. 20B, the beginning of the filter element deployment process, two filter elements 2000 and 2001 extend, vertically oriented, across the right hand portion of the air intake, having moved from a storage location (not shown in this figure) at the extreme right of the figure.

FIG. 20C shows further progression of the deployment process: Filter elements 2000 and 2001 have moved further to the left than their respective positions in FIG. 20B, and additional filter elements 2002 and 2003 have begun to traverse the intake opening.

FIG. 20D shows still further progression of the deployment process, beyond that shown in FIG. 20C: Filter elements 2000, 2001, 2002 and 2003 have moved still further to the left than their respective positions in FIG. 20C, and additional filter elements 2004 and 2005 have begun to traverse the intake opening.

FIG. 20E shows still further progression of the deployment process, beyond that shown in FIG. 20D: Filter elements 2000, 2001, 2002, 2003, 2004 and 2005 have moved still further to the left than their respective positions in FIG. 20D, and additional filter elements 2006 and 2007 have begun to traverse the intake opening.

15

As with the apparatus of FIGS. 19A to 19E, the exemplary set of FIGS. 20A to 20E shows a total of eight filter elements. Embodiments of the invention with a greater or lesser number of elements is possible. In the figure, in the deployed state, spacing between the elements is seen to be roughly equal. Embodiments of the invention in which the spacing is not equal are possible. The four step deployment process shown by FIGS. 20A to 20E is not intended to indicate that deployment is a step-wise process; it may be stepwise, or continuous; and if continuous, the elements may move from the retracted to the deployed position at a constant or non-constant speed.

FIGS. 20F-20J show a schematic view of an example of the process of deployment of both a vertically-oriented (2032) and horizontally-oriented (2034) filter element for engine 2030. FIG. 20F shows the fully retracted state; FIGS. 20G through 20I show gradual deployment and FIG. 20J shows full deployment. This mechanism can accommodate larger numbers of filter elements, and can accommodate filter elements with three or more orientations when fully deployed.

The relative width of the filter elements in the figures is not intended to indicate an actual relative width. The width may vary from element to element among an array of such elements. The width may vary along the length of an individual filter element. The vertical orientation of the filter elements is exemplary, and embodiment of the invention with horizontally oriented elements are possible, as well as embodiments in which the elements are neither vertical nor horizontal. Furthermore, embodiments of the invention are possible in which the filter elements are not parallel to each other—either during the process of deployment or in the fully deployed state.

FIG. 21 shows tandem filter arrays 2100 and 2102 at the front end of engine 2104. The respective elements of 2100 and 2102 are oriented perpendicular to each other, but in other embodiments, may be at any angle. Embodiments of the invention with one or more additional arrays of filter elements (i.e. a third, fourth . . . array) are possible. The arrays need not be identical.

FIGS. 22A and 22B show the placement of guiding elements for the respective filter elements in a retracted configuration and in a partially deployed configuration respectively.

Referring first to FIG. 22B, a partially deployed array of filter elements is shown. The deployed elements include 2200 and 2210 (and two others); the non-deployed elements include 2208 and 2209. Two guiding elements 2202 and 2207 are shown at each end of the line segment which represents filter element 2200. These two guiding elements move 2200 between its deployed position (partially deployed position shown in FIG. 22B) and its retracted position (shown in FIG. 22A).

As discussed hereinbelow (in conjunction with FIGS. 25A and 25B), the guiding elements have structure which facilitates the deployment and take-up of the respective filter element. And as discussed hereinbelow (in conjunction with FIGS. 23A-C and 24A-C) the guiding elements may move either passively (i.e. caused to move by another structure) or may have active means of locomotion.

As shown in FIG. 22B, the guiding elements move between an inner housing 2218 and an outer housing 2216. Also shown in FIG. 22B are filter element 2208 with respective guiding elements 2204 and 2205 in a minimally deployed position; partially deployed guiding element pair 2203 and 2206; partially deployed filter element 2210 with a respective pair of guiding elements 2212 and 2214; and minimally deployed filter element 2209.

16

FIG. 22A (in which the filter and guiding elements correspond to those with identical element numbers to those in FIG. 22B) shows the retracted state of the filter array. Guiding element pair 2204 and 2205 are shown in the most leftwards position of the figure, with a visible remnant of the filter element extending between them. Guiding element 2203 abuts 2204, and its associated second guiding element 2206 abuts guiding element 2205. Guiding element 2202 abuts 2203, and its associated second guiding element 2207 abuts guiding element 2206. Guiding element pair 2212 and 2214 are shown on the right side of the figure (along with two other pairs of guiding elements), all in the retracted position.

FIGS. 22A and 22B are intended to be exemplary. Those skilled in the art will note many other possible embodiments and configurations, including those with a different number of filter elements; those in which all of the retracted elements are situated in a single group (e.g. on the left side of the figure); those in which retraction is not to the right or left side of the engine (e.g. top and bottom, or top only, or bottom only); those in which the housing configuration differs; those in which the housing shape is not circular; and those in which the guiding and filter elements are not restricted to a one-dimensional placement in the retracted state.

Apparatus corresponding to that shown in FIGS. 22A and 22B will accommodate the filter array shown schematically in FIGS. 20F-20J if the guiding elements for the horizontally oriented (when deployed) filter element(s) extend forward (schematically, above the plane of the figure) to a greater or lesser degree than the guiding elements of the vertically oriented (when deployed) filter element(s). This feature is necessary in filter arrays comprising filter elements with more than one orientation, to accommodate what would otherwise be a “crossing problem”—i.e. the intersection of filter elements if confined to a single plane. Thus the plane occupied by the filter elements of one orientation will differ from the plane occupied by filter elements of another orientation. Furthermore, the orientation of the aforementioned two planes need not be parallel, as long as no part of a filter element in one plane contacts a filter element of another plane, in the deployed state.

For filter arrays with three or more orientations of deployed filter elements, a similar increase in the number of guiding element protrusion amounts would be necessary.

FIGS. 23A-C show schematic representations of a mechanism for causing the movement of guiding elements 2300, 2302, 2304, 2306, 2308 and 2310 in a passive manner—i.e. the guiding elements themselves do not possess an active source of propulsion. The propulsion source during deployment is deployment motors 2312 and 2314. The propulsion source during retraction is retraction motor 2316. Linkage 2318 links guiding element 2300 to deployment motor 2312; linkage 2320 links guiding element 2300 to guiding element 2302; linkage 2322 links guiding element 2302 to 2304; and linkage 2324 links retraction motor 2316 to guiding element 2304.

During deployment, 2312 pulls 2300 via linkage 2318. Guiding element 2302 is caused to move by the motion of 2300, via linkage 2320; and guiding element 2304 is caused to move by the motion of 2302 via linkage element 2322. Linkage control elements 2326, 2328 and 2330 associated with each guiding element control the length of the linkage between guiding elements. Thus, during deployment these lengths increase, and during retraction they decrease. The linkage control elements, in a preferred embodiment of the invention will comprises a motor or motors.

During retraction, 2316 pulls 2304 via linkage 2324. Guiding element 2302 is caused to move by the motion of 2304, via

17

linkage **2322**; and guiding element **2300** is caused to move by the motion of **2302** via linkage element **2320**. During the process of retraction, the length of the linkage elements between each adjacent pair of guiding elements decreases, under control of linkage control elements **2326**, **2328** and **2330**. A variety of spring arrangements which will be obvious to those skilled in the mechanical arts may subsidize the retraction process.

The arrangement of linkage control elements need not be one per guiding element: For example, linkage control element(s) in guiding element **2300** could control (a) the length of the linkage between deployment motor **2312** and guiding element **2300** and (b) the length of the linkage between guiding element **2300** and guiding element **2302**. A similar dual function linkage control apparatus could be situated in association with guiding element **2304**. In the aforementioned arrangement, there would be no need for linkage control apparatus in guiding element **2302**.

The position of linkage motors in each of the drawings is not intended to indicate its relative position in the actual apparatus; such position will be obvious to those skilled in the art. The same is true of all of the other elements shown in the figures discussed hereinabove and hereinbelow.

The elements and their function in the lower right portion of the figure are analogous to those in the upper right portion. Furthermore, similar apparatus would be arranged on the left side of the apparatus. Such left sided apparatus could use deployment motors **2312** and **2314**, or a second set of deployment motors.

For the embodiment of the invention in which all of the filter elements are situated in one group in the retracted state, deployment motor(s) placement at the left side of the figure could be utilized.

The figure also shows a plurality of bearings, for example **2330A-C** for guiding element **2300**, to decrease friction during motion. Numerous possible arrangements of bearings are possible. A lubricating system to further decrease friction, not shown, will be desirable.

The figure also shows exemplary figure elements (e.g. **2332**) and housing **2334**.

FIG. **23B** is similar to FIG. **23A** except that two types of linkage elements are shown: linkage deployment elements **2338**, **2340** and **2342**, utilized during the deployment process, and linkage retraction elements **2350**, **2352** and **2354**. A single linkage control element (**2356**, **2358** and **2360**) is shown in conjunction with each guiding element; configurations with (a) twice as many such elements [i.e. one per linkage element], (b) a larger number of linkage control elements and a smaller number of such elements are possible.

FIG. **23C** is analogous to FIGS. **23A** and **23B**, except that in the configuration shown in FIG. **23C** each guiding element (for example **2370**) is associated with (a) a linkage element (e.g. **2372**) which links it directly to deployment motor **2376**, and (b) a linkage element (e.g. **2374**) which links it directly to retraction motor **2378**. In such a configuration, linkage control elements are not necessary; the deployment motor(s) and the retraction motor(s) perform this function.

FIG. **24A-24C** comprise apparatus analogous to that of FIGS. **23A-C**, but in **24A-C** the means for causing propulsion of the guiding elements is part of the guiding element, rather than a separate element.

FIG. **24A** shows a guiding element propulsion system in which each guiding element is propelled by a motorized apparatus (hereinabove and hereinbelow referred to as "motor"), indicated by hexagon shapes in figure contained in the guiding element apparatus. Filter element **2401** is shown to extend from guiding element **2400** with motor **2404** to

18

guiding element **2402** with motor **2414**. Additional motors are shown (elements **2406**, **2408**, **2410** and **2412**, one each in association with each of the shown guiding elements. The engine housing is indicated by **2420**.

FIG. **24B** is similar to **24A** except:

only the end guiding elements **2434** and **2436** have a respective motor (**2430** and **2432**); linkage elements **2438** and **2440** allow for the passive deployment of respective guiding elements **2446** and **2448** powered by motor **2430**; and linkage elements **2444** and **2442** allow for the passive deployment of respective guiding elements **2450** and **2452** powered by motor **2432**.

Examples of two types of linkage control formats and elements are shown in FIG. **24B**.

In the first example, two linkage control elements are shown in association with guiding element **2446**: Linkage control element **2456** is for slack control and tension maintenance of linkage element **2438**; and linkage control element **2458** is for slack control and tension maintenance of linkage element **2440**.

In the second example, one linkage control element is shown in association with guiding element **2450**: Linkage control element **2460** is for slack control and tension maintenance of each of linkage elements **2442** and **2444**.

Numerous other slack/tension maintenance arrangements are possible with zero to two linkage control elements in association with each guiding element.

The retraction process in FIG. **24B** could be powered:

By having motor **2430** cause a pushing action of guiding element **2434** in the direction of **2446**, and thence in the direction of **2448** (with a pulling motion away from these respective guiding elements have cause the deployment of these guiding elements); and by having motor **2432** cause a pushing action of guiding element **2436** in the direction of **2450**, and thence in the direction of **2452** (with a pulling motion away from these respective guiding elements have cause the deployment of these guiding elements);

By motors in the linkage control elements. For example motors **2456** and **2458** would cause guiding elements **2430** and **2448** to approach guiding element **2446**, and motor **2460** would cause guiding elements **2436** and **2452** to approach guiding element **2450**; In such an example, at least one additional linkage control motor **2464** would be required to complete the retraction process. (**2464** would cause the group of guiding elements **2434-2446-2448** to approach the group of guiding elements **2436-2450-2452**.)

FIG. **24C** shows another configuration for propulsion of guiding elements, in which

"pulling motors" **2470** and **2472** cause deployment of the filter array, with each causing respective guiding elements **2474** and **2480** to move leftwards in the figure; The motion of **2474** during deployment passively causes the motion of guiding elements **2476** and **2478**, while the motion of **2480** during deployment passively causes the motion of guiding elements **2482** and **2484**; and

"pulling motors" **2486** and **2488** cause retraction of the filter array, with each causing respective guiding elements **2478** and **2484** to move rightwards in the figure; The motion of **2478** during retraction passively causes the motion of guiding elements **2476** and **2474**, while the motion of **2484** during retraction passively causes the motion of guiding elements **2482** and **2480**.

Linkage control elements **248S**, **2490**, **2492** and **2494** manage slack/tension control. Numerous other slack/tension control configurations are possible.

19

In addition, the management of deployment and retraction for filter element configurations in which the retracted state includes two or more groupings of filter elements (as shown for example in FIG. 19) can be accomplished utilizing each of the concepts and mechanisms shown and discussed in conjunction with FIGS. 23A-C and 24A-C.

Numerous other configurations for causing the movement of guiding elements and the management of linkages between them will be apparent to those skilled in the art.

FIG. 25A shows one mechanism for allowing the length of a central segment 2500 of a deployed filter element to increase during the deployment process and decrease during the retraction process. The entire filter element consists of (a) a first end segment 2502 shown coiled around shaft 2506, (b) central segment 2500, and (c) a second end segment 2504 shown coiled around shaft 2508. The control of shafts 2506 and 2508 may be either active (motor), passive (spring) or a combination of the two. In the case of a spring-based source of torque for the take-up of slack amounts of the filter element during retraction, the energy for release of additional amounts of filter element during deployment would be caused by the motion of the pair of respective guiding elements.

FIG. 25B shows another configuration for making an increasing length of central segment available during filter array deployment, and stowage during filter array retraction. The filter element in the figure comprises:

a first end segment comprising 2520A (wound around shaft 2530) and unwound portion 2520B, a central segment 2522, and a second end segment comprising unwound portion 2524B and wound portion 2524A (around shaft 2532).

Exemplary bearings 2540 and 2542 are shown, intended to indicate a plurality of bearings with appropriate lubrication.

Whereas the shaft and supporting components of the apparatus shown in FIG. 25A are located approximately in the plane of the front of the engine, these items are located further toward the back of the engine in the FIG. 25B configuration.

Numerous other configurations for release and take-up of filter element material will be apparent to those skilled in the art.

Cleaning apparatus 2550 and 2552 are shown in proximity to the end segments. The cleaning apparatus removes, minimizes and/or consolidates debris that may become attached to the filter elements. It may comprise a nozzle and reservoir for application of a cleaning fluid to the filter element; it may comprise a nozzle and air compression device to apply a blast of air to the filter element; it may comprise warming apparatus to melt ice that may have accumulated on the filter element; it may comprise one or more brushes to mechanically clean the element.

In a preferred embodiment, a catchment apparatus will store accumulated debris. The apparatus may alternatively, or in addition be situated to clean the central segment of the filter element.

FIG. 26 shows an embodiment of the invention in which the array 2604 of filter elements 2602 rotates about the long axis of the engine 2600, to increase the efficiency of bird and debris deflection. Dotted lines in the figure indicate that the filter array is attached to the engine. Embodiments of the invention in which two tandem filter arrays (e.g. as shown in FIG. 21) both utilize such rotation are possible. In a preferred embodiment with tandem rotating filter arrays, the angular velocity and direction of rotation of the arrays would differ.

There has thus been shown and described novel retractable bird and debris deflector for an aircraft jet engine which fulfills all the objects and advantages sought therefor. Many changes, modifications, variations and other uses and appli-

20

cations of the subject invention will, however, become apparent to those skilled in the art after considering this specification and the accompanying drawings which disclose the preferred embodiments thereof. All such changes, modifications, variations and other uses and applications which do not depart from the spirit and scope of the invention are deemed to be covered by the invention, which is to be limited only by the claims which follow.

What is claimed is:

1. A retractable bird and debris deflector for an aircraft jet engine comprising an air intake duct with a central longitudinal axis and a forward opening for receipt of air, said forward opening having a perimeter and said deflector comprising, in combination:

(A) a plurality of elongate members disposed on the air intake duct of the engine in spaced relation to each other, each member having two end segments and a central segment disposed between the two end segments;

(B) a plurality of guiding members, each mounted for movement along a perimeter of the intake duct and coupled to one end segment of an elongate member, wherein the central segment of each elongate member extends between a pair of guiding members such that the elongate member is movable by said pair of guiding members between a retracted position and a deployed position in front of the intake duct such that:

(a) in a first, deployed position, said central segments are arranged in front of said forward opening, in spaced relation; and

(b) in a second, retracted position, said central segments are removed from the forward opening of said intake duct;

whereby said central segments of said elongate members, when in said deployed position, are situated to impede the ingress of debris into said intake duct.

2. The deflector apparatus defined in claim 1, wherein said central segments are maintained in a substantially parallel relationship in the retracted position.

3. The deflector apparatus defined in claim 1, wherein said central segments are maintained in a substantially parallel relationship in the deployed position.

4. The deflector apparatus defined in claim 1, wherein said elongate members are arranged in a single closely spaced grouping when in said retracted position.

5. The deflector apparatus defined in claim 1, wherein said elongate members are arranged in a plurality of closely spaced groupings when in said retracted position.

6. The deflector apparatus defined in claim 1, wherein each said elongate member comprises a flexible cable.

7. The deflector apparatus defined in claim 6, wherein each said cable is comprised primarily of metal.

8. The deflector apparatus defined in claim 6, wherein each said cable is stretchable.

9. The deflector apparatus defined in claim 1, wherein said end segments of said elongate members are disposed inside a housing of said intake duct.

10. The deflector apparatus of claim 6, further comprising a cable take-up mechanism for selectively pulling and releasing said cable forming each elongate member.

11. The deflector apparatus defined in claim 6, further comprising a plurality of motors, each coupled to a respective elongate member, for maintaining tension in said cable during movement from said deployed position to said retracted position.

12. The deflector apparatus defined in claim 11, wherein each said motor is operable to release a respective cable while

21

maintaining cable tension during movement from said retracted position to said deployed position.

13. The deflector apparatus of claim 1, further comprising a motorized mechanism for selectively moving said guiding members.

14. The deflector apparatus defined in claim 1, further comprising at least one motor, coupled to at least one of said pair of guiding members for moving said respective guiding member along said perimeter of said intake duct during movement from said deployed position to said retracted position.

15. The deflector apparatus defined in claim 14, further comprising at least one second motor, coupled to another one of said pair of guiding members for moving said other respective guiding member along said perimeter of said intake duct during movement from said deployed position to said retracted position.

16. The deflector apparatus defined in claim 1, further comprising at least one motor, coupled to one of said pair of guiding members for moving said respective guiding member along said perimeter of said intake duct during movement from said retracted position to said deployed position.

17. The deflector apparatus defined in claim 16, further comprising at least one second motor, coupled to another one of said pair of guiding members for moving said respective guiding member along said perimeter of said intake duct during movement from said retracted position to said deployed position.

18. The deflector apparatus as defined in claim 1 forming a first deflector apparatus, said deflector apparatus further comprising a second deflector apparatus comprising:

(C) a plurality of second elongate members disposed on the air intake duct of the engine in spaced relation to each other, each second member having two end segments and a central segment disposed between the two end segments;

(D) a plurality of second guiding members, each mounted for movement along a perimeter of the intake duct and coupled to one end segment of a second elongate member, wherein the central segment of each second elongate member extends between a pair of second guiding mem-

22

bers such that the second elongate member is movable by said pair of said second guiding members between a retracted position and a deployed position in front of the intake duct, said central segments of said second elongate members being oriented, in said deployed position, substantially in parallel to each other but not in parallel to the elongate members of said first deflector apparatus, thereby to more fully protect said engine from the ingress of debris.

19. The deflector apparatus defined in claim 18, wherein said central segments of said second elongate members of said second deflector apparatus are oriented substantially perpendicular to said central segments of said elongate members of said first deflector apparatus.

20. The apparatus defined in claim 18, wherein a plane defined by the central segments of said elongate members is different than a plane defined by the central segments of said second elongate members; thereby to prevent intersection between the elongate members and the second elongate members.

21. The deflector apparatus defined in claim 1, wherein said elongate members rotate about an axis perpendicular to a longitudinal axis of said jet engine.

22. The deflector apparatus defined in claim 1, further comprising a cleaning apparatus, coupled to at least one of said guiding members for cleaning said elongate members.

23. The apparatus defined in claim 22, wherein said cleaning apparatus is selected from the group consisting of:

(i) a nozzle, a reservoir coupled to said nozzle for supplying a cleaning fluid, and a pump coupled to said nozzle for dispersing said fluid through said nozzle, for providing a pressurized spray of said fluid for cleaning said elongate members;

(ii) a nozzle, an air intake for supplying air, and a pump coupled to said nozzle for providing pressurized air for cleaning said elongate members;

(iii) a brush for cleaning said elongate members; and

(iv) a heating element, coupled to a power supply for heating said element, for applying heat to said and thereby cleaning said elongate members.

* * * * *